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2-DIMENSIONAL AXISYMMETRIC AND 3- DIMENSIONAL FINITE
ELEMENT STRESS ANALYSIS OF THE LHA-1 CLASS SUPERHEATER
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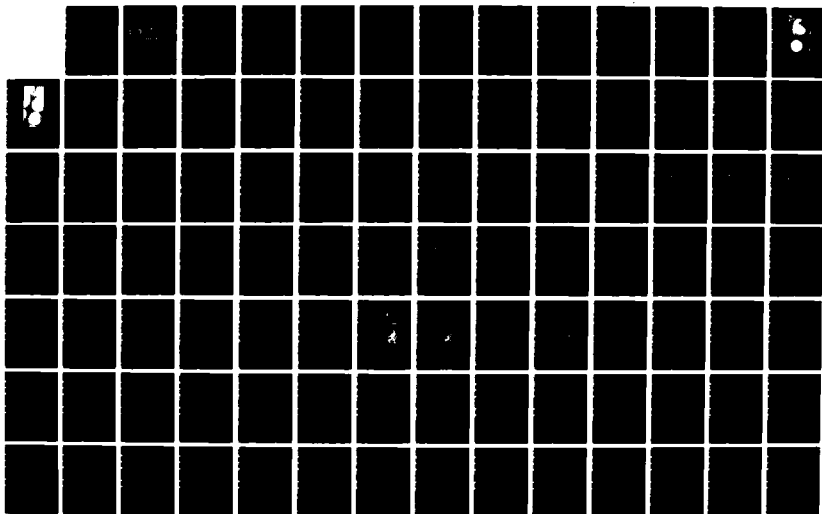
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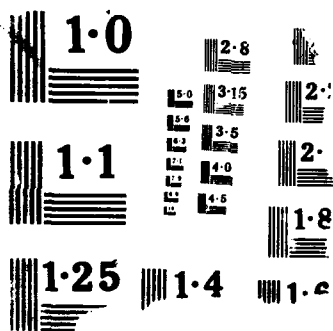
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THESIS

2-DIMENSIONAL AXISYMMETRIC AND
3-DIMENSIONAL FINITE ELEMENT
STRESS ANALYSIS OF THE LHA-1 CLASS SUPER-
HEATER HEADER

by

Doyle R. Kitchin

March 1988

Thesis Advisor

G. Cantin

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2-dimensional axisymmetric and 3-dimensional finite element
stress analysis of the LHA-1 class superheater header

by

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Lieutenant Commander, United States Navy
B.S.M.E., United States Naval Academy, 1976

Submitted in partial fulfillment of the
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ABSTRACT

ADINA, a finite element program for automatic dynamic incremental nonlinear analysis, is used to conduct a stress analysis of the Tarawa class (LHA-1) superheater header. This investigation is sponsored by the Naval Sea Systems Command (NAVSEA) as a result of cracking in the superheater header tube attachment weld. A 2-dimensional solid axisymmetric finite element model was developed using both a coarse and fine mesh. Both pressure and thermally induced stresses were studied. A 3-dimensional solid finite element model was developed and a preliminary stress analysis conducted.

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I. INTRODUCTION

The Tarawa (LHA-1) class amphibious assault ships have been experiencing numerous failures of their superheater header. The failures occur in the tube attachment weld. This is a relatively new class of ship and they are expected to be in commission for another twenty to thirty years. The solution of this problem is imperative to prevent the excessive down time experienced by these engineering plants.

A 2-dimensional solid axisymmetric finite element model was developed using ADINA. A study of the effect of different thermal gradients across the thickness of the header was conducted. The model yielded very different results dependent on the boundary conditions imposed. A discussion is made of the boundary conditions and the conclusion that a 3-dimensional model would more closely model the actual physical boundary conditions of this structure.

A 3-dimensional solid finite element model was developed. A discussion is made of the boundary conditions and a study is made of the internal pressure loading.

II. DESCRIPTION OF THE PROBLEM

NAVSEA has been conducting an extensive investigation to determine the cause for the weld joint cracking. The following list was provided by NAVSEA and is an analysis of weld joint cracking on the USS Tarawa (LHA-1):

- Linear defects were located by MT inspection in original fabrication welds and repair welds alike, including welds installed as recently as 1982.
- Predominant linear defect was the transverse weld crack, occurring singly and in multiples of two or more; crack indications ranged from 1 1/16 inch to 5 8/16 inch in length.
- Numerous linear indications were also found across the thin ligaments between welds in the A-to-D loop direction; cracks occurred singly and in multiples, and ranged in length from 1 1/16 inch to 3 1/4 inch.
- Predominant crack orientation was transverse to the longitudinal axis of the header; very few axially oriented indications were observed.
- Cracked welds in the inlet/outlet header were concentrated in the first and fourth passes.
- No apparent concentration of defects in the welds adjacent to the division plates (region of highest thermal gradient).
- The inlet/outlet header examined contained nearly 2-1/2 times as many weld joints with linear defects as did the intermediate header (32% of welds in no. 1 boiler inlet/outlet header vs. 13% of welds in no. 2 boiler intermediate header).
- Destructive analysis indicated the maximum crack depth to be approximately 5/32 inch, and the crack propagation mechanism to be fatigue at elevated temperature.

Figure 1 on page 3 and Figure 2 on page 4 are photographs taken by NAVSEA showing magnetic particle indications of the cracking. A more detailed description of this problem is given in [Ref. 1: pp. 10-15].



Figure 1. Magnetic particle indications of transverse weld cracks in no. 2 boiler intermediate header tube



Figure 2. Magnetic particle indications of ligament cracks in no. 1 boiler inlet/outlet header tube attachment joints

III. DESCRIPTION OF ADINA

ADINA is a finite element program for automatic dynamic incremental nonlinear analysis. The ADINA system is composed of four main programs: ADINA-IN, ADINA, ADINA-PLOT and ADINAT. The 1984 version was used for this thesis. The sequence of execution of these programs is shown in Figure 3 on page 6 which is taken from [Ref. 2: p. 1-2].

A. GENERATION OF INPUT DATA

ADINA-IN generates the input data for ADINA or ADINAT. ADINA is used for displacement and stress analysis and ADINAT performs the temperature analysis. ADINA-IN can be run in an interactive or batch mode.

Coordinates can be input in a local or global coordinate system using cartesian, cylindrical or spherical coordinates. There are several automatic mesh generation methods available.

ADINA-IN has the ability to graphically check the model. All or part of the model can be drawn from any view desired, including element and nodal numbering. The LINES parameter described in [Ref. 2: p. 5.25-15] gives some ability to specify which lines on the model will be drawn. This allows an outline of the model to be drawn and the ability to check for missing elements.

The procedure for utilizing ADINA-IN to generate input data for ADINA on the VAX/VMS operating system is explained in Appendix A. The procedure for utilizing ADINA-IN to generate input data to solve the thermally induced stress problem is explained in Appendix B. The reverse Cuthill-McKee bandwidth optimization is used in generating the input file for ADINA.

B. POST-PROCESSING

ADINA-PLOT performs the post-processing of the output data from ADINA. It also has graphics capability, allowing a view of both the deformed and undeformed shapes. The deformed scale factor can be specified. ADINA-PLOT can also do vector plotting of the principal stresses.

Several listing options are available. Displacements or stresses for all or part of the model can be printed. Maximum values or results exceeding a specified value can also be listed.

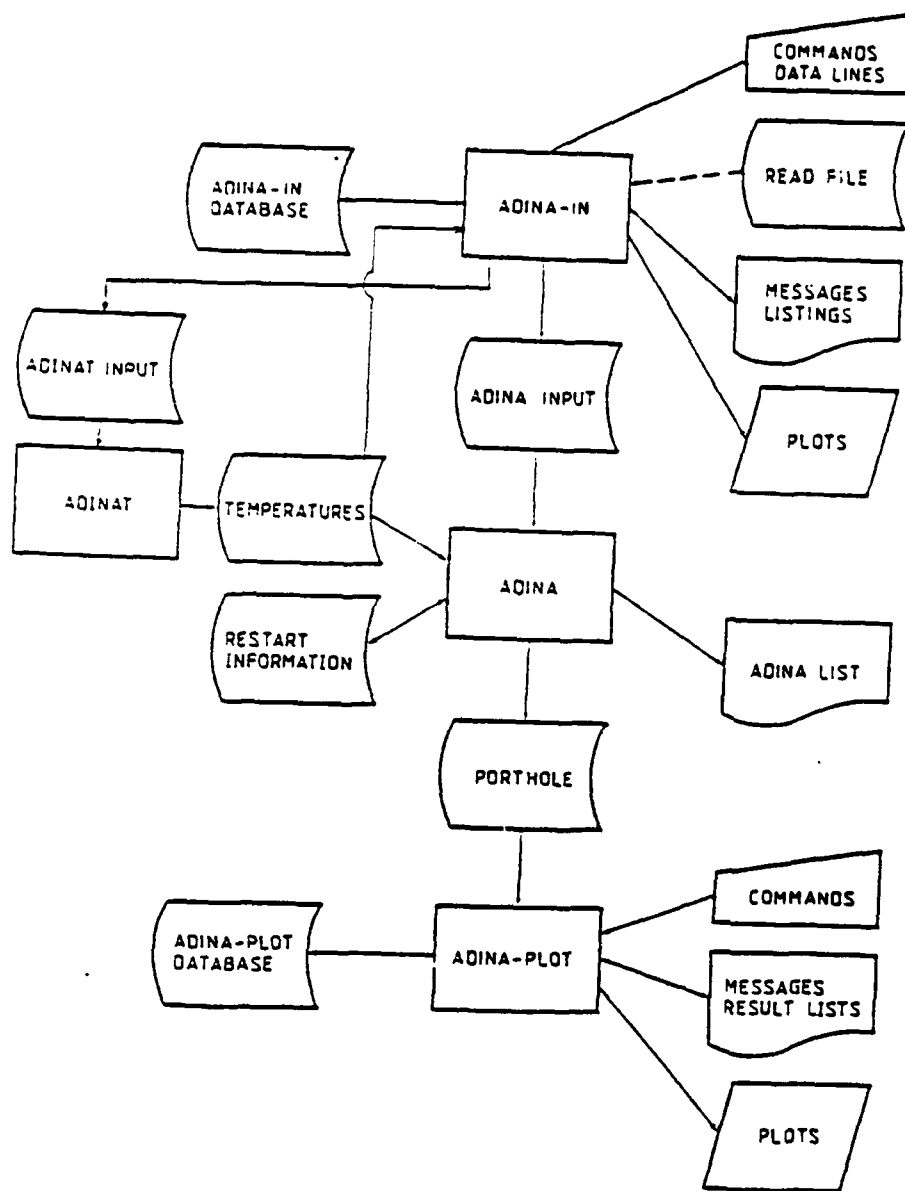


Figure 3. Sequence of execution of ADINA programs

The procedure for operating ADINA-PLOT on the VAX/VMS operating system is explained in Appendix C.

C. METHOD OF SOLUTION

ADINA uses isoparametric displacement based finite elements. The basic assumptions from [Ref. 3: p. 2.3-8] are:

for the coordinates

$$x = \sum_{i=1}^q h_i x_i \quad y = \sum_{i=1}^q h_i y_i \quad z = \sum_{i=1}^q h_i z_i \quad (3.1)$$

for the displacements

$$u = \sum_{i=1}^q h_i u_i \quad v = \sum_{i=1}^q h_i v_i \quad w = \sum_{i=1}^q h_i w_i \quad (3.2)$$

where:

- $h_i(r,s,t)$ = the interpolation function corresponding to node i
- r,s,t = the isoparametric coordinates
- q = the number of element nodes
- x_i, y_i, z_i = nodal point coordinates
- u_i, v_i, w_i = nodal point displacements

ADINA uses the principle of virtual displacement to generate the equilibrium equations. The equilibrium equation as listed in [Ref. 4: p. 124] is:

$$K U = R \quad (3.3)$$

where:

- K = the element stiffness matrix
- U = the unknown nodal point displacements
- R = the load vector which includes body forces, surface forces, initial stresses and concentrated loads.

The development of the stiffness matrix for the isoparametric element is given in [Ref. 4: pp. 203-204]:

$$K = \int_V B^T C B dV \quad (3.4)$$

where:

- B = the strain-displacement matrix
- C = the elasticity matrix

In terms of the isoparametric coordinates:

$$dV = \det(J) dr ds dt \quad (3.5)$$

where:

- $\det(J)$ = the determinant of the Jacobian operator

The stresses are calculated using the strains at the point of interest. The basic relationships used are given in [Ref. 4: p. 123]:

$$\epsilon = B U \quad (3.6)$$

$$\tau = C \epsilon \quad (3.7)$$

where:

- ϵ = the element strain matrix
- τ = the element stress matrix

The element matrices are calculated using Gauss numerical integration. A (3x3x3) integration order is utilized as specified in [Ref. 3 : p. 2.3-16].

D. AXISYMMETRIC PROBLEM

The basic assumptions for the axisymmetric element are listed in [Ref. 3: p. 2.2-4]:

$$\epsilon_{xx} = u/y \quad (x \text{ is the hoop direction}) \quad (3.8)$$

$$\tau_{xy} = \tau_{xz} = 0 \quad (y \text{ is the radial direction}) \quad (3.9)$$

If the loading is also axisymmetric, a 2-dimensional analysis of a unit radian of the structure will provide the complete stress and strain distributions. This greatly reduces the computational time compared to a 3-dimensional analysis.

E. CONTACT SURFACES

ADINA allows the specification of contact surfaces in the development of the model. Contact surfaces are defined in [Ref. 3: p. 4.1-1] as surfaces that are initially in contact, or are anticipated to come into contact during the response solution.

The contact surface problem is a nonlinear analysis. After each equilibrium iteration, the segment contact force is evaluated from the nodal forces. A contactor segment is shown in Figure 4 on page 10. The conditions of either tension release, sliding or sticking are determined for each segment. The basic equations are given in [Ref. 3: p. 4.2-3]. Tension release occurs if:

$$\vec{\eta} \cdot \vec{T}_n^j < 0 \quad (3.10)$$

where:

- $\vec{\eta}$ = the inward normal vector to segment j as shown in Figure 4 on page 10
- \vec{T}_n^j = the normal contact force for segment j

Tension release means the surfaces are not in contact.

If the surfaces are in contact the condition of sliding or sticking is determined by using the friction law. Sticking occurs if:

$$\vec{T}_t^j < \mu \vec{T}_n^j \quad (3.11)$$

where:

- \vec{T}_t^j = the tangential contact force for segment j
- μ = the coefficient of friction

Sliding occurs if:

$$\vec{T}_t^j > \mu \vec{T}_n^j \quad (3.12)$$

The nodal forces are calculated as the consistent nodal loads which correspond to the tangential and normal forces of the segment.

F. THERMAL ANALYSIS

ADINAT uses isoparametric finite element discretization. The basic assumptions from [Ref. 5: p. 11] are:

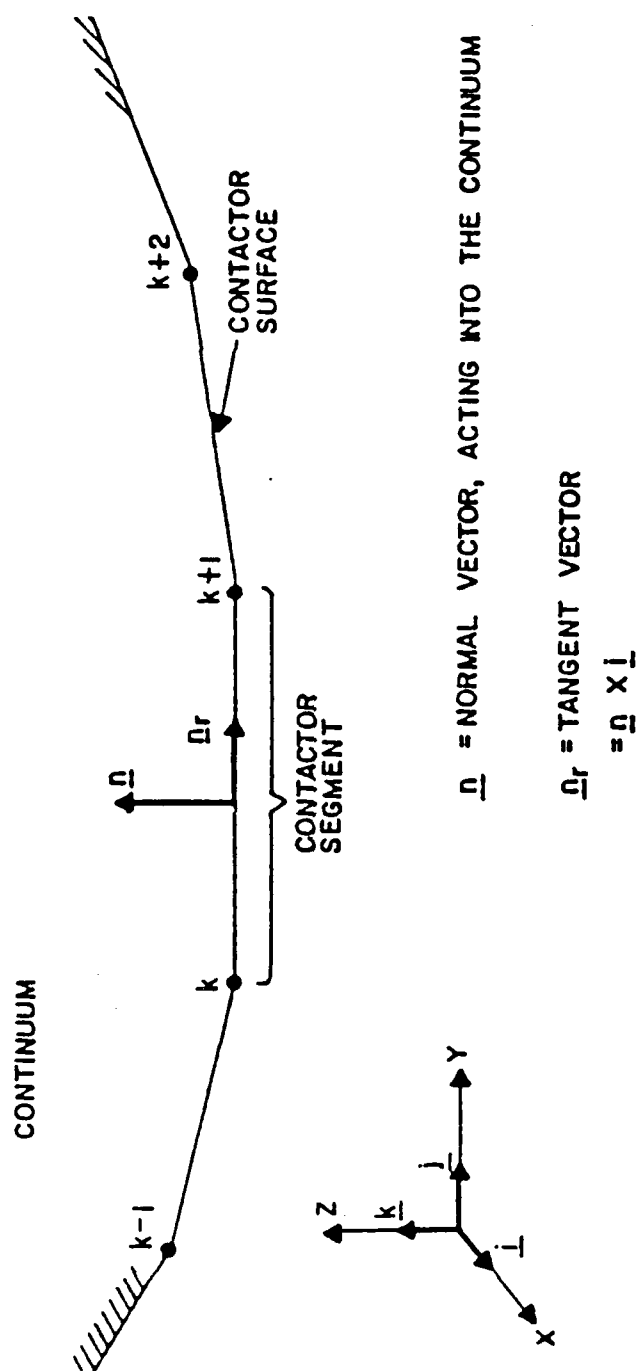


Figure 4. Contact surface contactor segment

$$x = \sum_{i=1}^N h_i x_i \quad y = \sum_{i=1}^N h_i y_i \quad z = \sum_{i=1}^N h_i z_i \quad (3.13)$$

and:

$$t + \Delta t_\theta = \sum_{i=1}^N h_i t + \Delta t_{\theta i} \quad t_\theta = \sum_{i=1}^n h_i t_{\theta i} \quad \Delta \theta = \sum_{i=1}^n h_i \Delta \theta_i \quad (3.14)$$

where:

- h_i = the element interpolation function
- N = the number of nodal points
- x_i, y_i, z_i = coordinates of the nodal point i
- $t + \Delta t_{\theta i}$ = the temperature of nodal point i at time $t + \Delta t$
- $t_{\theta i}$ = the temperature of nodal point i at time t
- $\Delta \theta_i$ = the temperature increment at nodal point i

The interpolation and substitution of these equations into the general incremental heat flow equilibrium equations, yield the finite element equations.

G. VERIFICATION

Part of the installation process of a finite element code is the verification of the code. This is done by modeling a problem for which the results are known and comparing the results.

Three verifications are included in this thesis. The solution of a cantilever beam problem is compared with classical beam theory in Appendix D. The thermal stresses in an annular disk are compared with classical disk theory in Appendix E. The temperature distribution in a chimney is compared with a verified finite difference solution in Appendix F. The last two problems were chosen because of their similarity to the problem addressed in this thesis. In each case ADINA yielded excellent results.

IV. 2-DIMENSIONAL SOLID AXISYMMETRIC FINITE ELEMENT MODEL

This problem was originally modeled by Lieutenant Jon W. Kaufmann, United States Navy, in [Ref. 1]. The initial attempt of this thesis was to duplicate his model.

A. INITIAL MODEL DEVELOPMENT

NAVSEA provided a cross section drawing of the superheater header as illustrated in Figure 5 on page 13. They also provided a detail of the superheater header tube attachment weld joint in Figure 6 on page 14. The same geometry of the weld as developed in [Ref. 1: p. 18] was utilized and is illustrated in Figure 7 on page 15.

A nine-node plane axisymmetric element was used. This element is similar to the one used in [Ref. 1: p. 29].

A coarse and fine mesh were developed in order to do a study on convergence. The coarse mesh has 64 elements and 292 nodes. The model is shown in Figure 8 on page 16. A close-up of the weld is shown in Figure 9 on page 17, listing element numbers. The fine mesh has 224 elements and 1095 nodes and is illustrated in Figure 10 on page 18. The weld area is shown in Figure 11 on page 19.

The same boundary conditions stated in [Ref. 1: p. 30] were applied. The right hand side of the model was fixed as illustrated in Figure 12 on page 20. A frictional surface was applied between the tube and the header. The nodes on the boundary were constrained together in the y direction but allowed to move with respect to each other in the z direction. This is in effect applying a contact surface with the coefficient of friction equal to zero.

B. LOAD CONDITIONS

The same load conditions were applied as developed in [Ref. 1: pp. 38-42].

1. Load condition no. 1 - Internal pressure load

Load condition no. 1 consisted of an internal pressure of 700 psi applied at all nodes located on the inside of the tube and header. This corresponds to the full rating of the boiler. This loading is illustrated in Figure 13 on page 21.

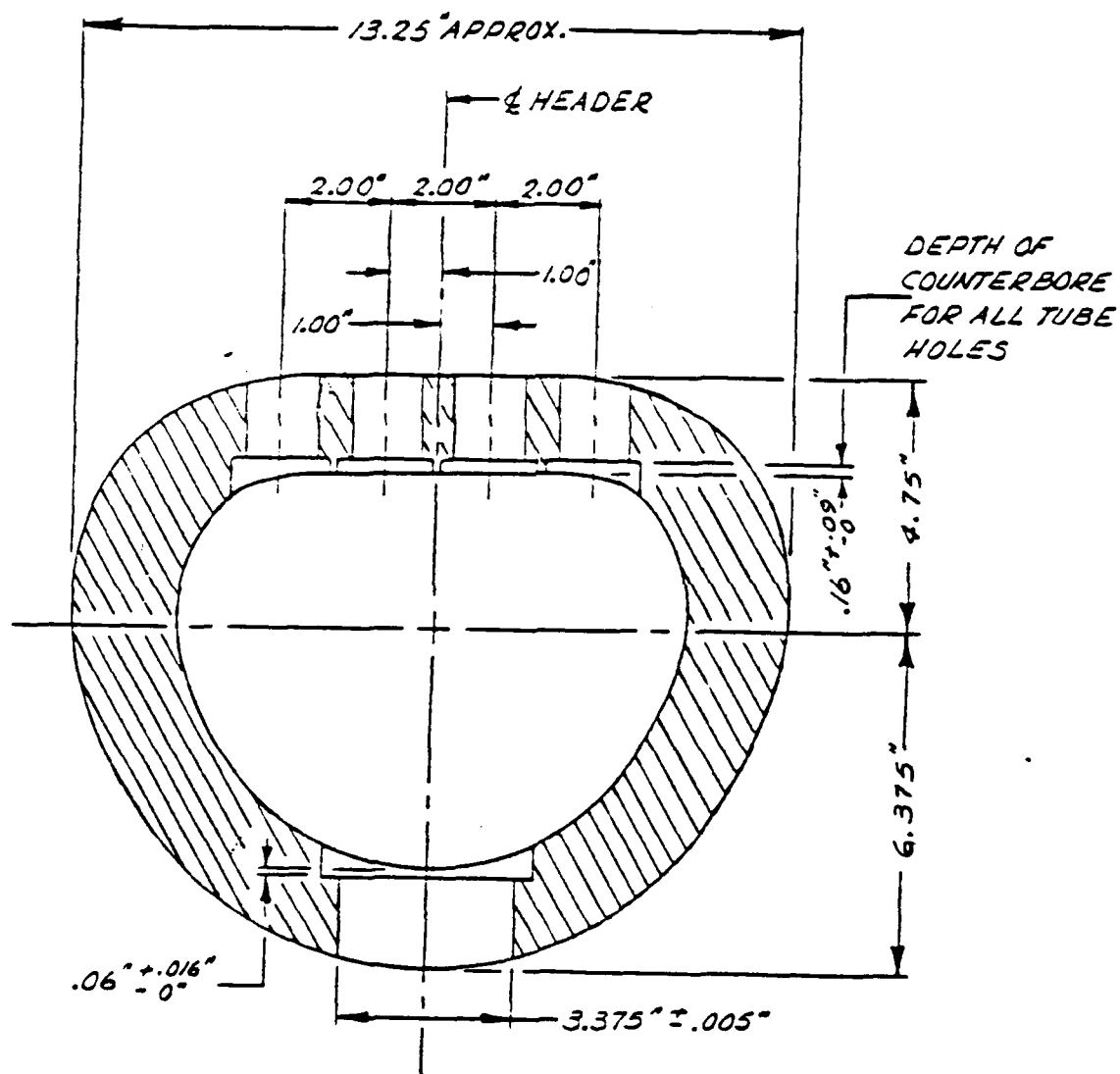
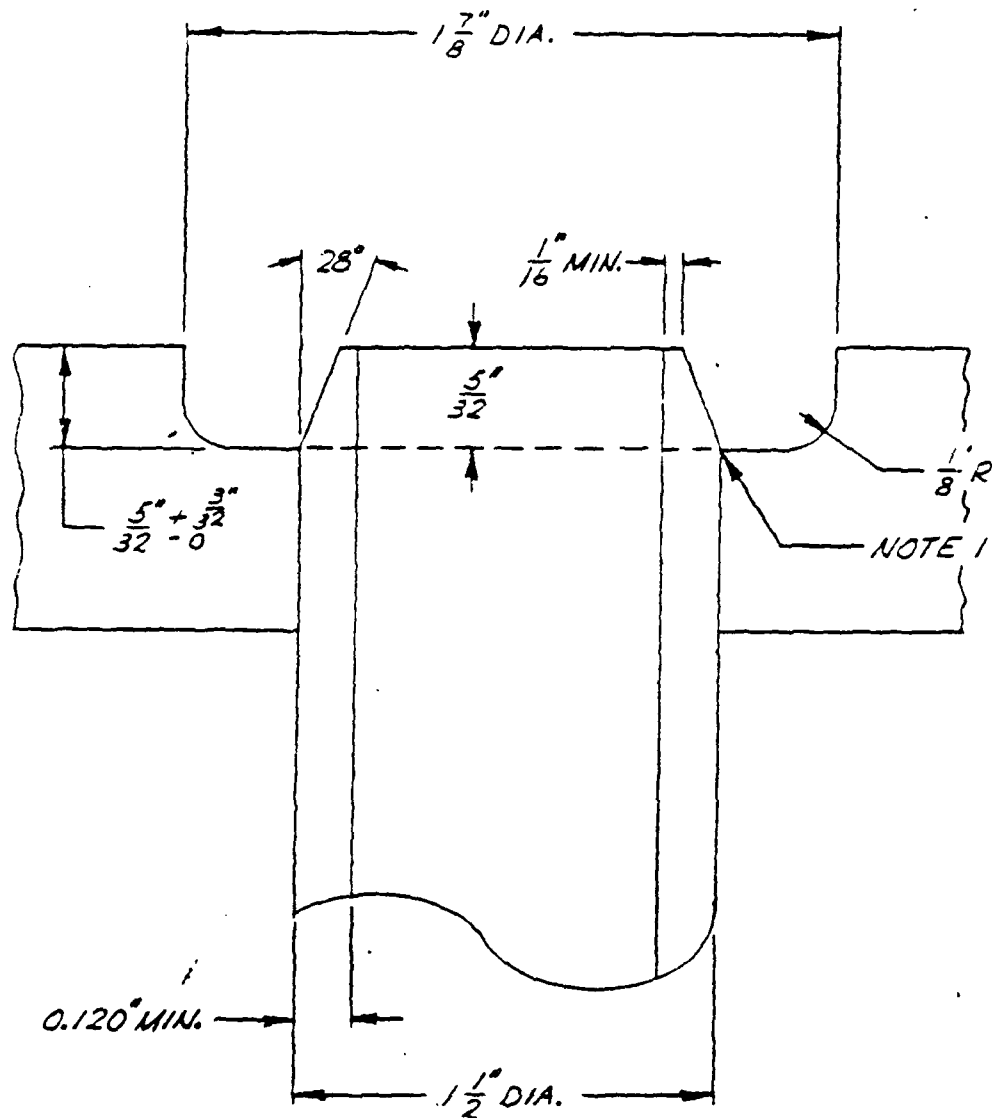


Figure 5. Cross sectional view of the superheater header



NOTE:

1. TUBE IS POSITIONED SUCH THAT BASE OF TUBE CHAMFER IS FLUSH WITH SQUARE CORNER AT BOTTOM OF COUNTERBORE AS SHOWN

Figure 6. Detail of the attachment weld joint

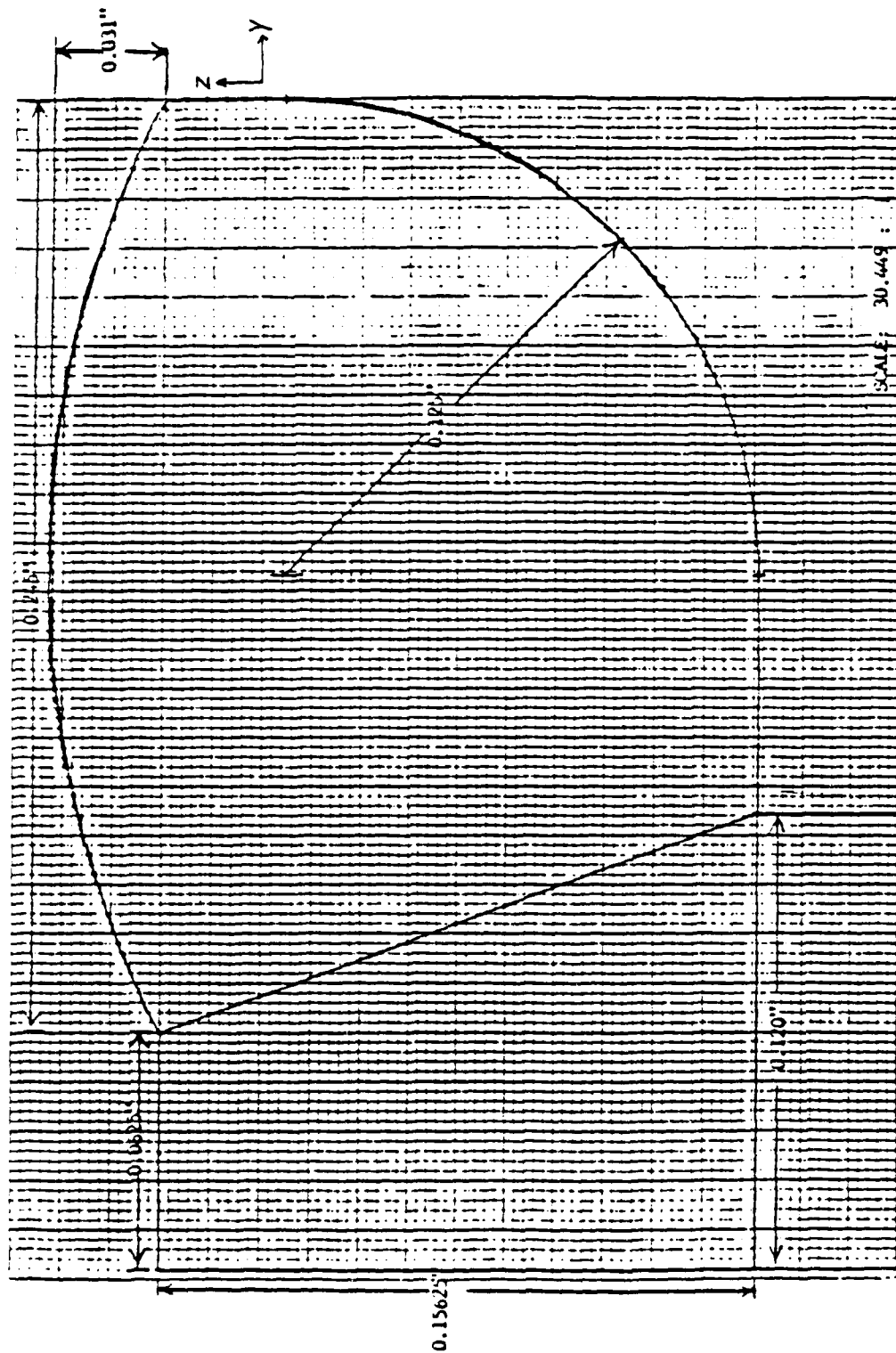


Figure 7. Geometry of the weld

HEADER
ORIGINAL ——— D.262

Z
└── Y

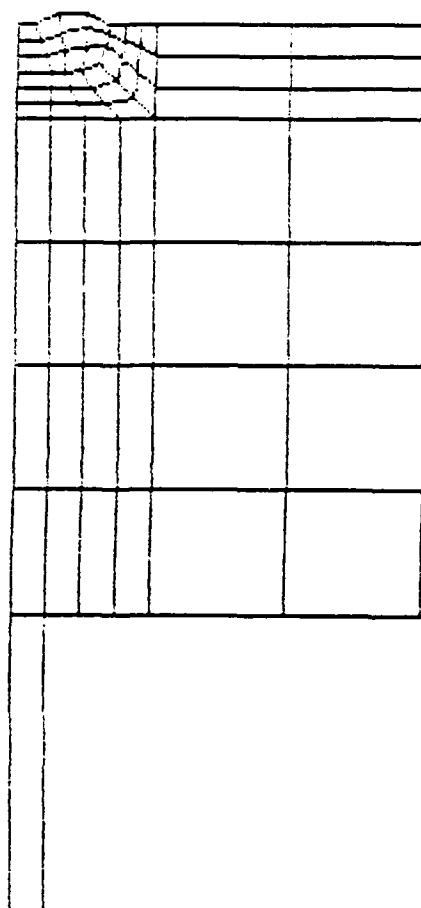


Figure 8. Coarse mesh

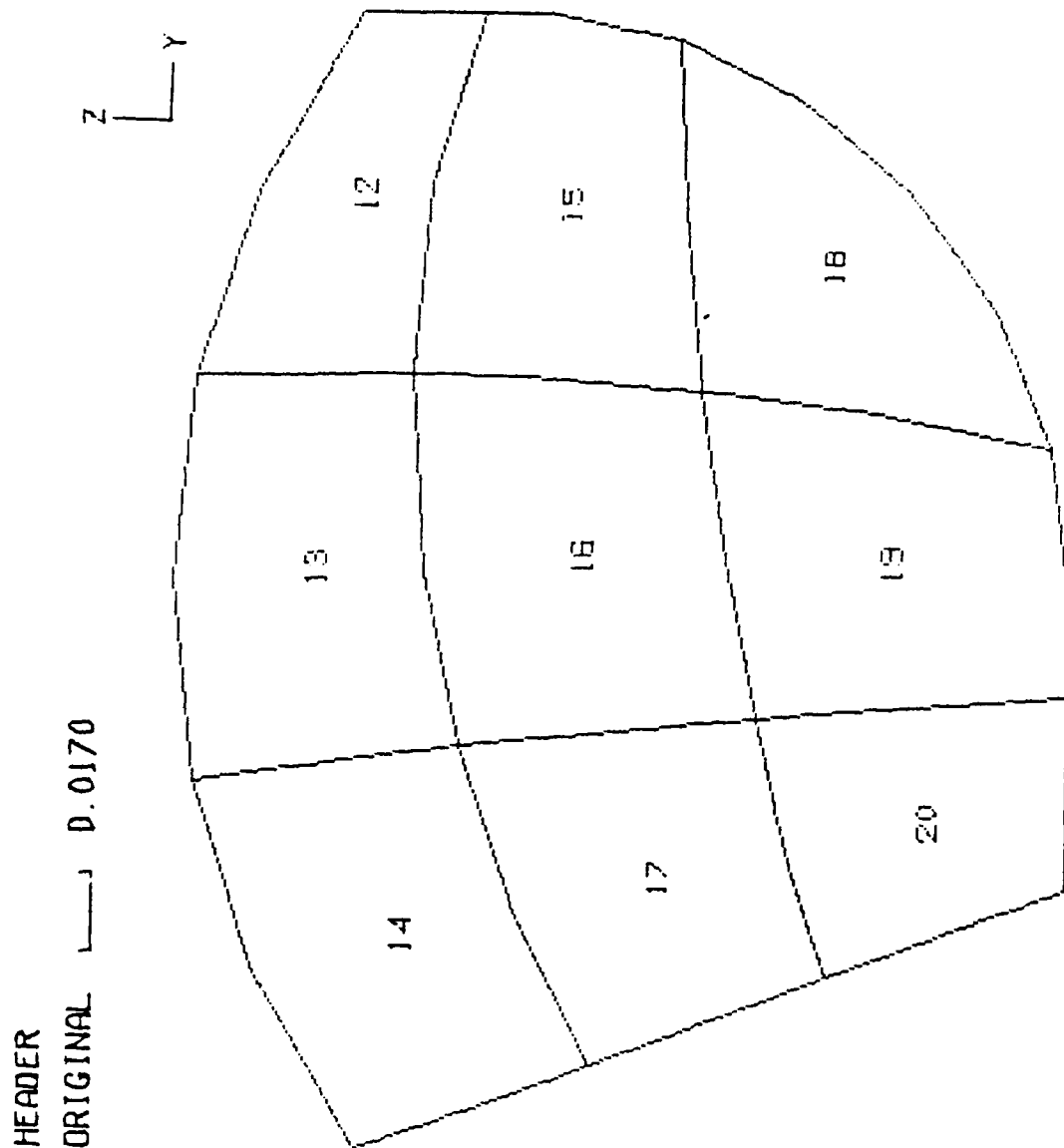



Figure 9. Close-up of the weld area, coarse mesh

HEADER
ORIGINAL  D.252

Z
Y

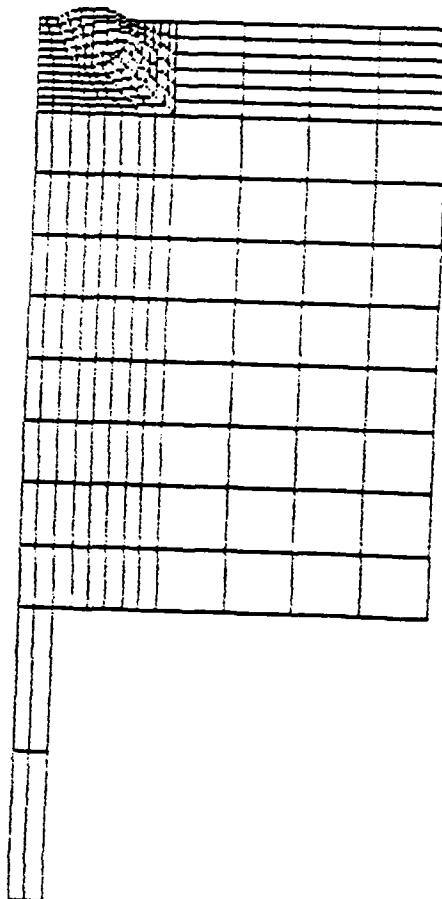


Figure 10. Fine mesh

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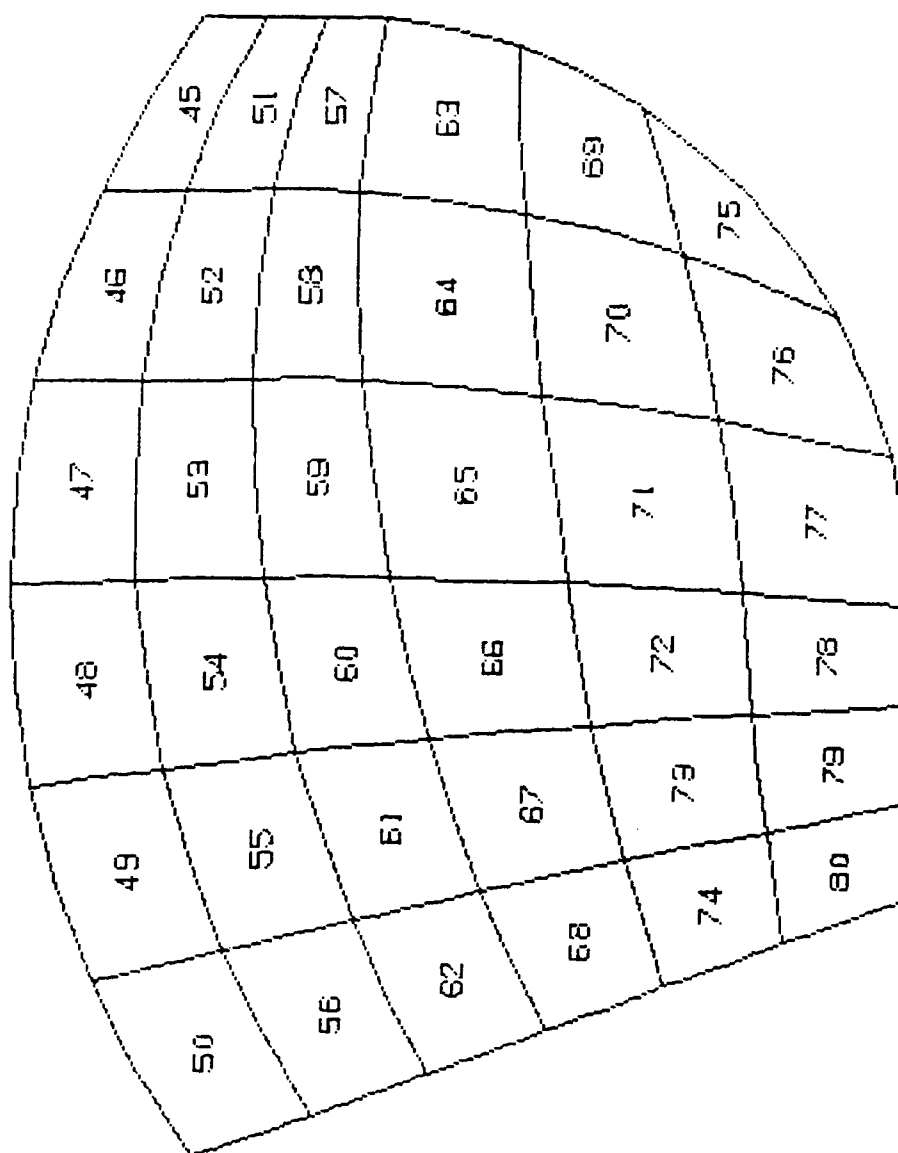



Figure 11. Close-up of the weld area, fine mesh

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Z
Y

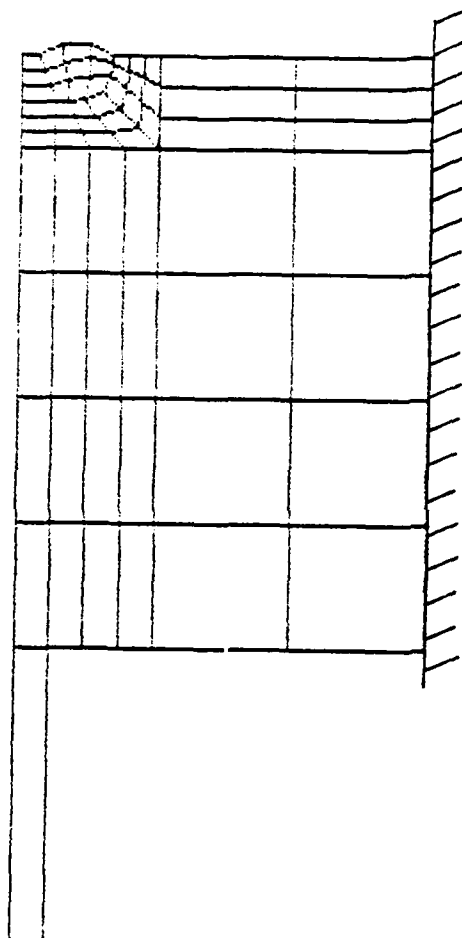


Figure 12. Fixed boundary conditions

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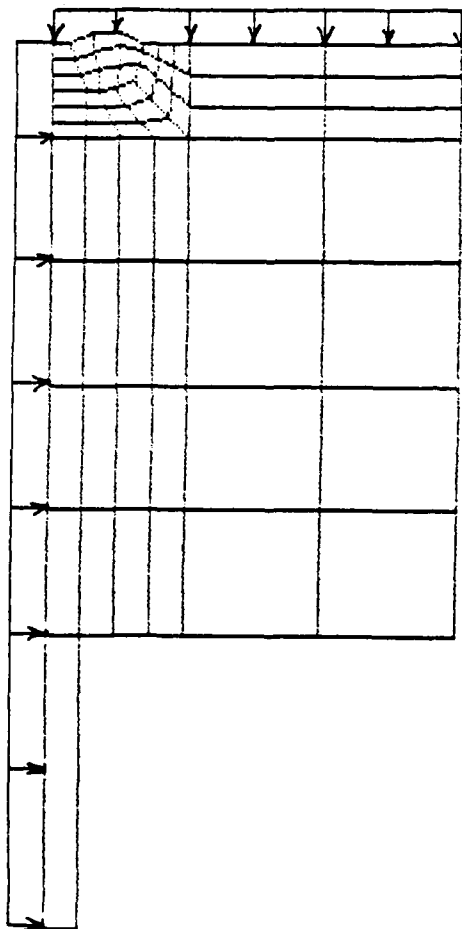
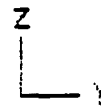


Figure 13. Load condition no. 1

2. Load condition no. 2 - Longitudinal tube load

The force per radian is 152 lbf. The coarse model uses the same consistent loading as [Ref. 1: p. 40]. The load was applied to the bottom element of the tube, on nodes 85, 68 and 87 as indicated in the following list:

- $F_{85} = F' (1/6) = 25.3 \text{ (Lb}_f\text{)}$
- $F_{68} = F' (2/3) = 101.3 \text{ (Lb}_f\text{)}$
- $F_{87} = F' (1/6) = 25.3 \text{ (Lb}_f\text{)}$

This is illustrated in Figure 14 on page 23. The consistent loading for the fine mesh was applied to nodes 85, 240, 239, 238 and 87 as follows:

- $F_{85} = F' (1/12) = 12.66666666 \text{ (Lb}_f\text{)}$
- $F_{240} = F' (1/3) = 50.66666666 \text{ (Lb}_f\text{)}$
- $F_{239} = F' (1/6) = 25.33333333 \text{ (Lb}_f\text{)}$
- $F_{238} = F' (1/3) = 50.66666666 \text{ (Lb}_f\text{)}$
- $F_{87} = F' (1/12) = 12.66666666 \text{ (Lb}_f\text{)}$

The consistent loading for the fine mesh is illustrated in Figure 15 on page 24.

3. Load condition no. 3 - Thermal load

The thermal load consists of maintaining a constant temperature of 350 degrees F at the interior nodes of the tube and header. A temperature gradient of 100 degrees F is applied to the right hand side of the model. This load condition is shown in Figure 16 on page 25.

C. MATERIAL PROPERTIES

The same material properties were used as listed in [Ref. 1: p. 38]. Young's Modulus (E) equals 29,600,000 psi, the coefficient of thermal expansion (α) equals .0000065 in./in./ $^{\circ}\text{F}$ and Poisson's ratio (ν) equals .3. The material of the tube and header is 2.25% chromium, 1% molybdenum steel (ASTM grade 22). The weld was assumed to have the same properties as the tube and header.

D. COMPARISON OF RESULTS

The original and deformed shapes for load conditions 1, 2 and 3 for the coarse mesh are shown in Figure 17 on page 26, Figure 18 on page 27, and Figure 19 on page 28. The results for the fine mesh are shown in Figure 20 on page 29, Figure 21 on page 30, and Figure 22 on page 31. The maximum stresses in the weld for the coarse and fine

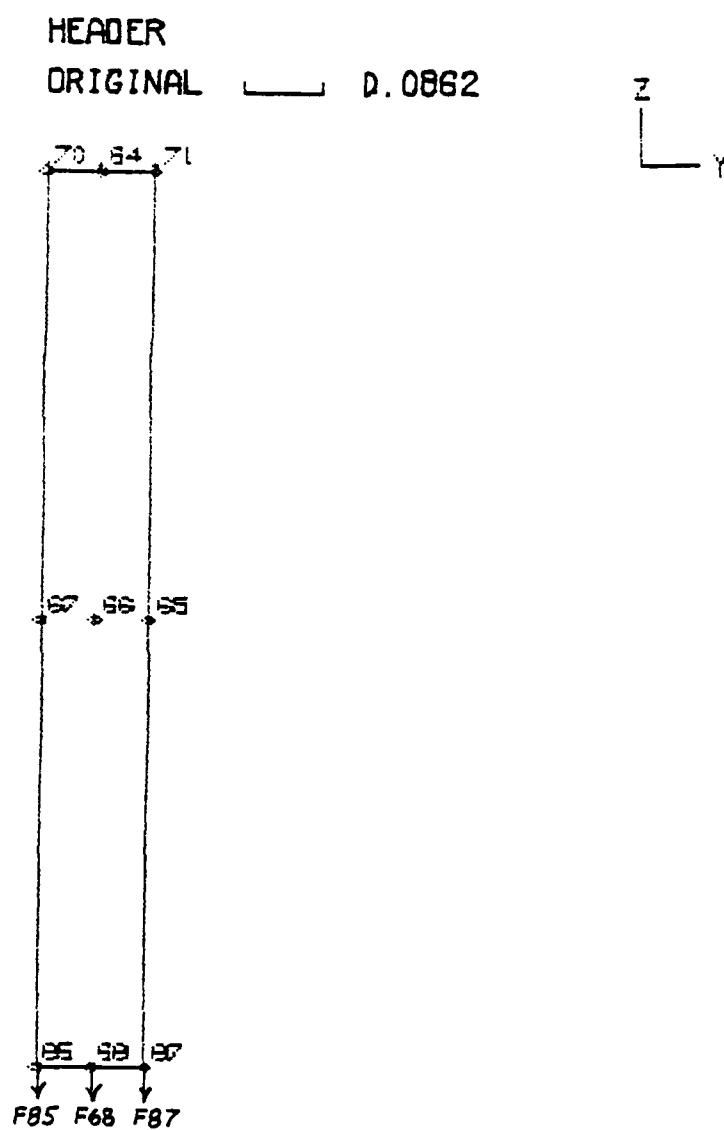


Figure 14. Load condition no. 2, coarse mesh

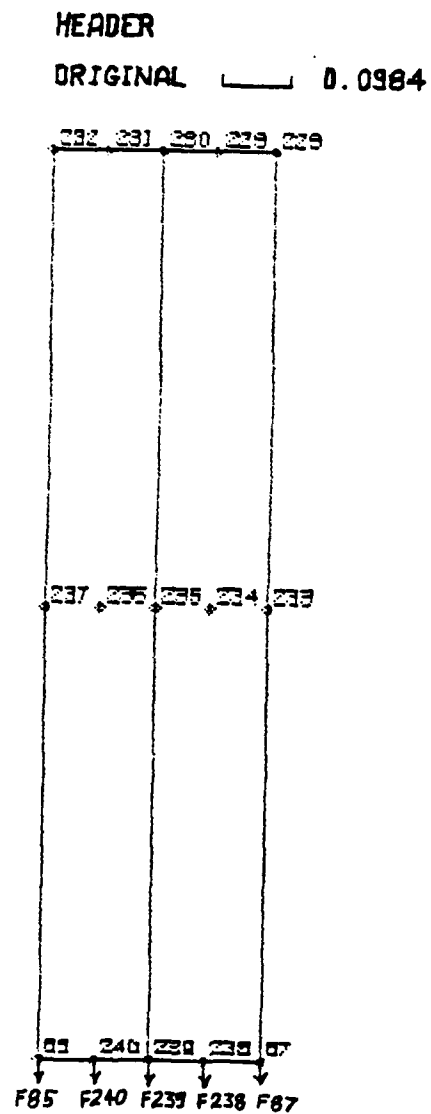


Figure 15. Load condition no. 2, fine mesh

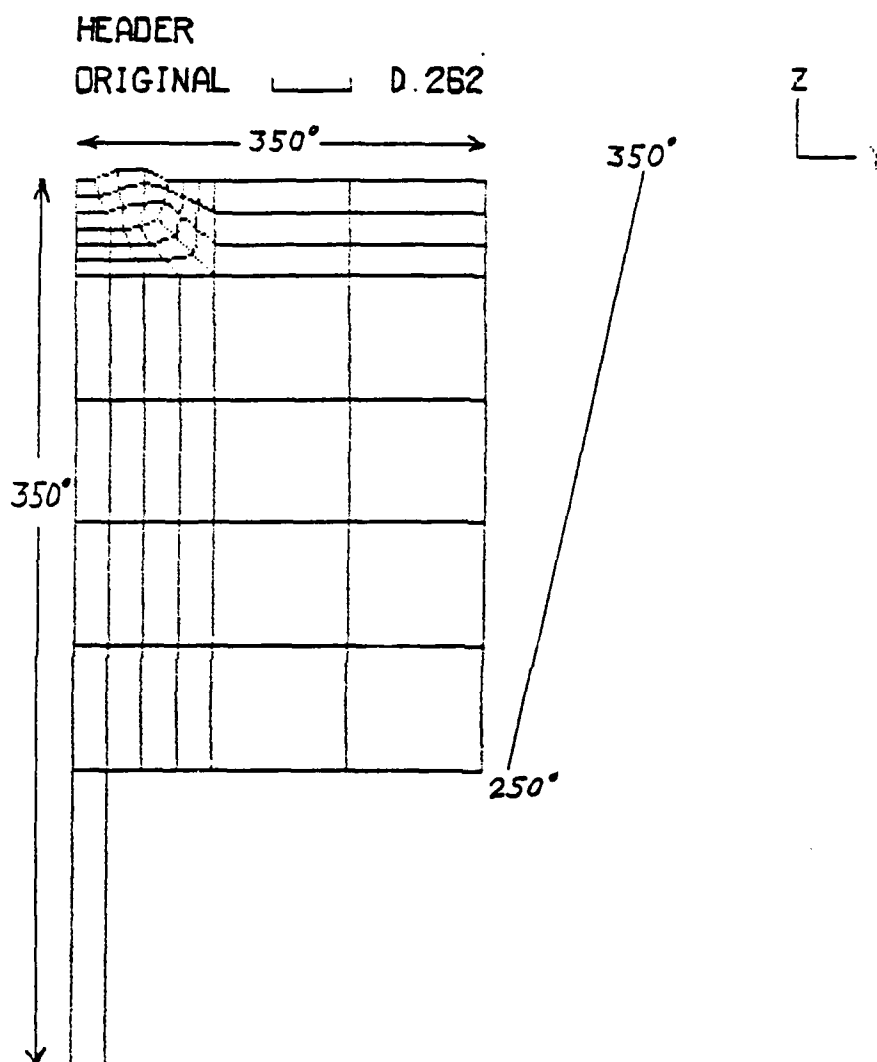


Figure 16. Load condition no. 3

HEADER

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DEFORMED L D.0847
TIME 0.000000100

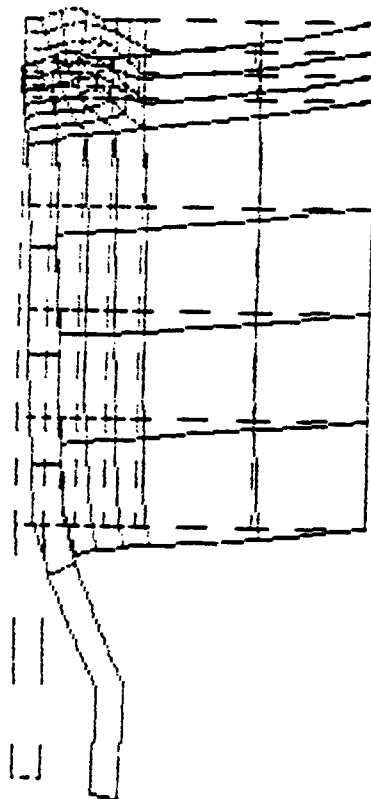
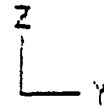



Figure 17. Original and deformed shape of coarse mesh for load condition no. 1

HEADER

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DEFORMED  D.172

TIME 0.000000100

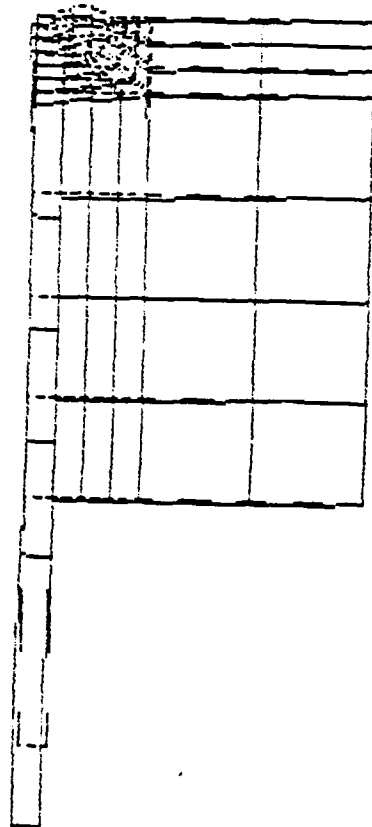
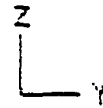



Figure 18. Original and deformed shape of coarse mesh for load condition no. 2

HEADER

ORIGINAL  D.343

DEFORMED  D.00606

TIME 0.000000100

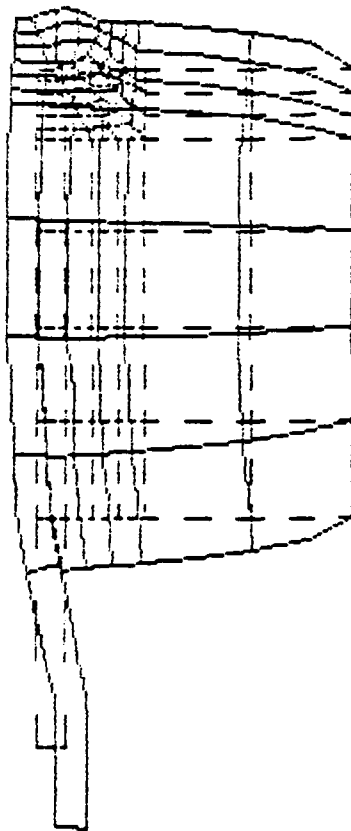
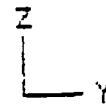




Figure 19. Original and deformed shape of coarse mesh for load condition no. 3

HEADER

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DEFORMED  D.0000871
TIME 2.000

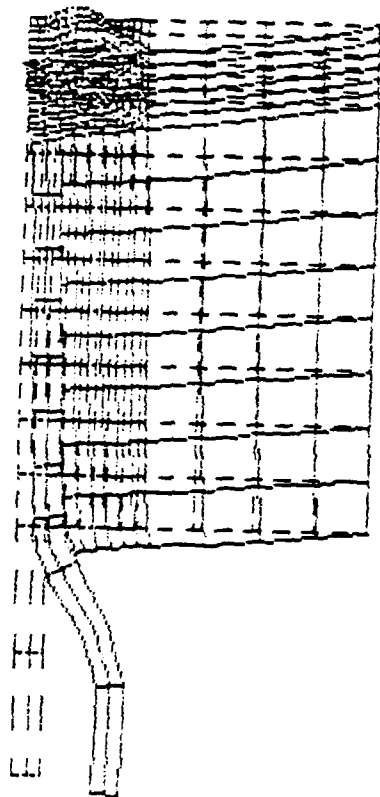
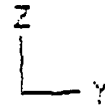


Figure 20. Original and deformed shape of fine mesh for load condition no. 1

HEADER

ORIGINAL  D.319

DEFORMED  D.000174

TIME 2.000

z
|
└─ y

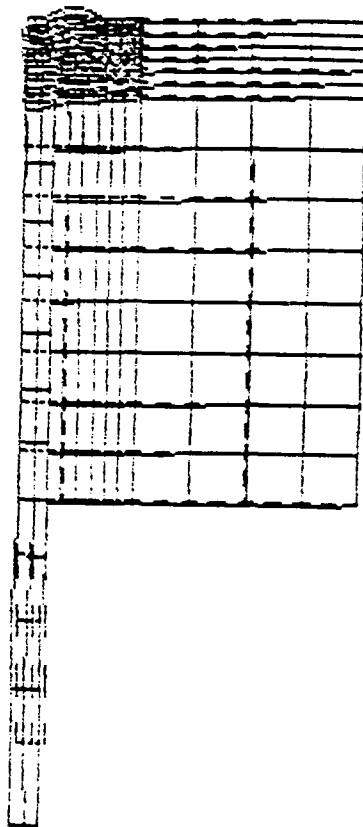




Figure 21. Original and deformed shape of fine mesh for load condition no. 2

HEADER

ORIGINAL  D.344
DEFORMED  D.00590
TIME 2.000

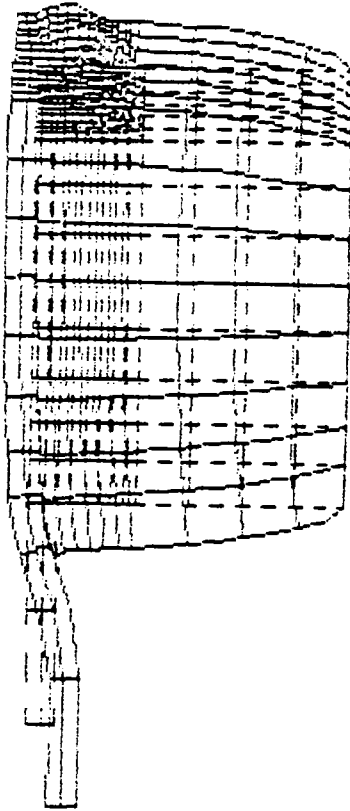
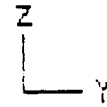


Figure 22. Original and deformed shape of fine mesh for load condition no. 3

mesh are listed in Table 1 and Table 2. For the axisymmetric problem, σ_{xx} is the hoop stress, σ_{yy} is the radial stress, and σ_{zz} is the longitudinal stress.

Table 1. MAXIMUM STRESSES FOR THE COARSE MESH

Stresses	Load Condition No. 1	Load Condition No. 2	Load Condition No. 3
σ_{xx} (psi)	-.636767E+03	-.509766E+03	-.135183E+06
σ_{yy} (psi)	-.661331E+03	-.370531E+03	-.466026E+05
σ_{zz} (psi)	-.742839E+03	-.116857E+03	-.559437E+04
σ_{xy} (psi)	.120933E+03	.332004E+03	.134600E+05

Table 2. MAXIMUM STRESSES FOR THE FINE MESH

Stresses	Load Condition No. 1	Load Condition No. 2	Load Condition No. 3
σ_{xx} (psi)	-.649740E+03	-.534142E+03	-.136960E+06
σ_{yy} (psi)	-.659697E+03	.503587E+03	-.555045E+05
σ_{zz} (psi)	-.753965E+03	-.145094E+03	-.980113E+04
σ_{xy} (psi)	.136478E+03	.372948E+03	.193084E+05

The magnitude and direction of the principal stresses for the coarse and fine mesh, in the area of the weld, are listed in Table 3 and Table 4. A vector plot of the principal stresses of the coarse and fine mesh for load condition no. 3 is shown in Figure 23 on page 34 and Figure 24 on page 35.

Table 3. PRINCIPAL STRESSES FOR THE COARSE MESH


Element	P1 (psi)	P2 (psi)	Angle ($^{\circ}$ F)
12	-.1594E+03	-.3422E+05	74.57
13	0.1352E+04	-.1820E+05	85.39
14	0.2802E+03	-.1413E+05	94.46
15	0.9153E+03	-.3779E+05	86.39
16	0.3717E+04	-.2807E+05	86.38
17	0.2304E+04	-.2084E+05	86.18
18	0.1716E+04	-.3808E+05	88.03
19	0.3323E+04	-.3259E+05	87.72
20	0.3103E+04	-.2530E+05	86.98

Table 4. PRINCIPAL STRESSES FOR THE FINE MESH

Element	P1 (psi)	P2 (psi)	Angle ($^{\circ}$ F)
45	-.8641E+03	-.4057E+05	67.64
46	-.8286E+03	-.2143E+05	73.66
47	-.8814E+03	-.1572E+05	82
48	-.6244E+03	-.1315E+05	90.31
49	-.2067E+03	-.1219E+05	98.49
50	-.5970E+03	-.1293E+05	103.33
51	-.1553E+04	-.4200E+05	80.67
52	-.2516E+04	-.3006E+05	79.02
53	-.2590E+04	-.2272E+05	82.48
54	-.2192E+04	-.1923E+05	86.37
55	-.1592E+04	-.1725E+05	89.20
56	-.2959E+03	-.1531E+05	88.85
57	-.2147E+04	-.4153E+05	85.36
58	-.2898E+04	-.3440E+05	84.10
59	-.3626E+04	-.2802E+05	84.60
60	-.3324E+04	-.2402E+05	85.88
61	-.2581E+04	-.2107E+05	86.41
62	-.1310E+04	-.1789E+05	86.05
63	-.1468E+04	-.4044E+05	87.31
64	-.2494E+04	-.3631E+05	87.32
65	-.3856E+04	-.3161E+05	86.70
66	-.3772E+04	-.2755E+05	86.46
67	-.3167E+04	-.2403E+05	86.19
68	-.2107E+04	-.2036E+05	85.89
69	-.1507E+02	-.3976E+05	87.96
70	-.2439E+04	-.3685E+05	88.22
71	-.3705E+04	-.3322E+05	87.71
72	-.3793E+04	-.2956E+05	87.17
73	-.3364E+04	-.2615E+05	86.76
74	-.2680E+04	-.2254E+05	86.44
75	-.1317E+04	-.3915E+05	88.01
76	-.2801E+04	-.3685E+05	88.09
77	-.3636E+04	-.3375E+05	97.90
78	-.3742E+04	-.3055E+05	87.62
79	-.3503E+04	-.2759E+05	87.37
80	-.3035E+04	-.2437E+05	87.13

The maximum hoop stress listed in [Ref. 1: pp. 39-40] for load condition no. 1 is .69 (ksi) and the maximum hoop stress for load condition no. 2 is .96 (ksi). Those results compare favorably with the results shown here from ADINA. [Ref. 1: p. 42] indicates the maximum hoop stress for load condition no. 3 is 19.0 (ksi). This value is much smaller than the result of 137 (ksi) obtained with ADINA. In considering the verification of ADINA as explained in chapter three, it is believed that the results obtained here are correct. The results confirm that the thermal stresses are the predominant stresses.

HEADER

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TIME 0.000000100

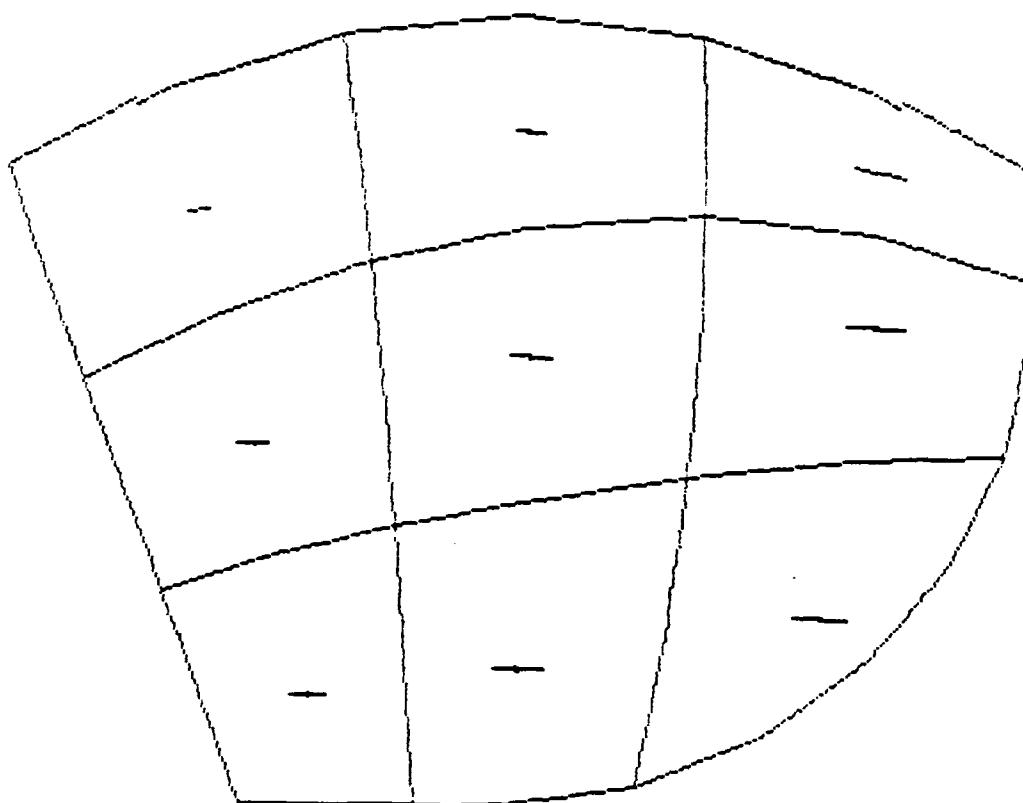
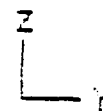



Figure 23. Vector plot of the principal stresses of the coarse mesh for load condition no. 3

HEADER
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TIME 2.000

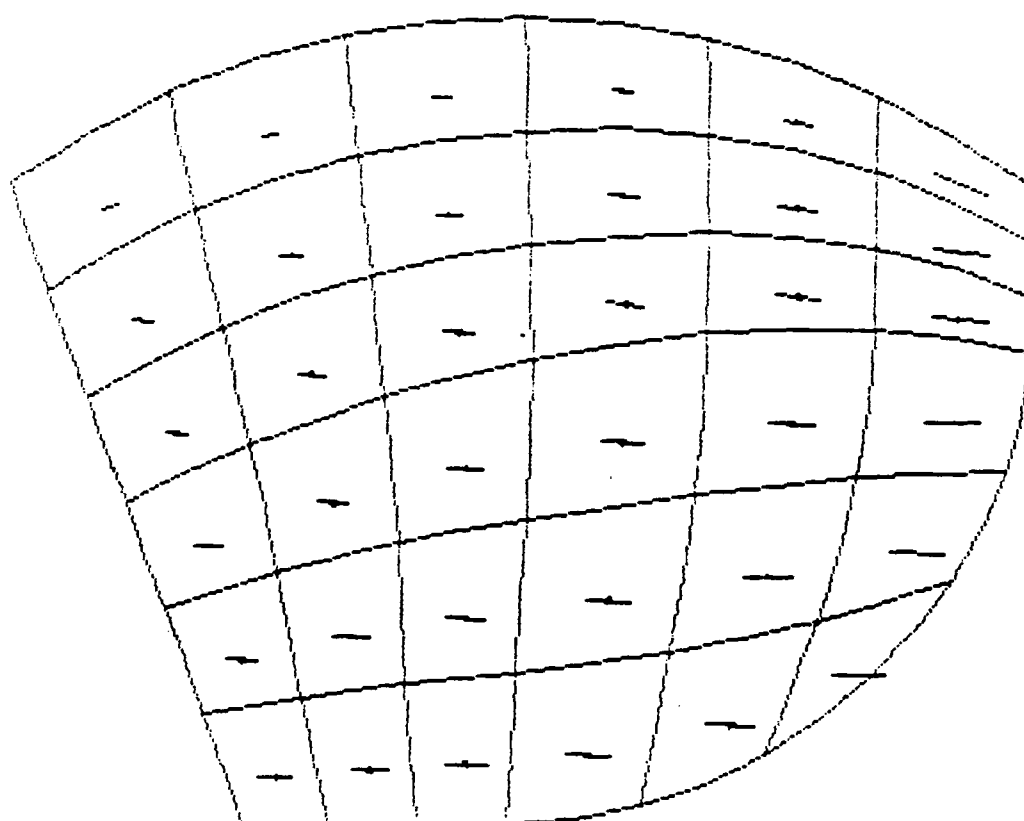
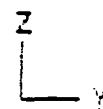


Figure 24. Vector plot of the principal stresses of the fine mesh for load condition no. 3

The accuracy of the results of the problem that has been modeled are very good. There is very little difference between the results of the coarse and fine mesh which means convergence has been achieved. There is no significant difference in the calculated stresses at the boundaries of adjacent elements.

The next step was to use the contact surface option of ADINA for the boundary between the tube and the header. This allows an indication of the coefficient of friction and a determination of its effect on the solution. The maximum stresses are shown in Table 5 and Table 6 for the coarse and fine mesh for a coefficient of friction of zero. These results compare very well with the previous results.

Table 5. MAXIMUM STRESSES FOR THE COARSE MESH, CONTACT SURFACE OPTION

Stresses	Load Condition No. 1	Load Condition No. 2	Load Condition No. 3
σ_{xx} (psi)	-.646923E+03	-.514228E+03	-.135265E+06
σ_{yy} (psi)	-.657614E+03	.362559E+03	-.465231E+05
σ_{zz} (psi)	-.747963E+03	-.112506E+03	-.558613E+04
σ_{xz} (psi)	.134382E+03	.328554E+03	.134376E+05

Table 6. MAXIMUM STRESSES FOR THE FINE MESH, CONTACT SURFACE OPTION

Stresses	Load Condition No. 1	Load Condition No. 2	Load Condition No. 3
σ_{xx} (psi)	-.646745E+03	-.537235E+03	-.137053E+06
σ_{yy} (psi)	-.657250E+03	.495395E+03	-.553930E+05
σ_{zz} (psi)	-.757047E+03	-.139909E+03	-.978366E+04
σ_{xz} (psi)	.139310E+03	.366954E+03	.192709E+05

E. FURTHER MODEL DEVELOPMENT

A detailed study of the problem and the results obtained in the previous section indicated some changes were needed in the model.

It was discovered that the model as developed was not a repeating section. The distance between the center of the tubes is two inches. The right hand side of the model is two inches from the center of the tube. A new model was developed with the right hand side being one inch from the center of the tube. The coarse mesh consists of 43

elements and 211 nodes. The fine mesh has 172 elements and 763 nodes. The models are shown in Figure 25 on page 38 and Figure 26 on page 39.

Figure 19 on page 28 and Figure 22 on page 31 prompted a change in boundary conditions. The deformed shapes showed that the header was not allowed to expand uniformly in the axial direction. The bottom node of the right hand side of the header was fixed and the rest of the nodes on the right hand side were allowed to roll in the z direction. These boundary conditions are shown in Figure 27 on page 40.

The original and deformed shapes for load conditions 1, 2 and 3 for the coarse mesh are shown in Figure 28 on page 41, Figure 29 on page 42 and Figure 30 on page 43. Figure 30 on page 43 shows that the header is now expanding uniformly in the axial direction. The maximum stresses are indicated in Table 7 and Table 8 for the coarse and fine mesh. The ADINA-IN input file for these models is listed in Appendix G.

Table 7. MAXIMUM STRESSES FOR THE COARSE MESH, REPEATING SECTION


Stresses	Load Condition No. 1	Load Conditions 1 and 2	Load Condition No. 3
σ_{xx} (psi)	-.420381E+03	-.566508E+03	-.997953E+05
σ_{yy} (psi)	-.597283E+03	.792176E+03	-.373304E+05
σ_{zz} (psi)	-.825134E+03	-.112852E+04	-.454326E+04
σ_{xz} (psi)	.231712E+03	.838120E+03	.104692E+05

Table 8. MAXIMUM STRESSES FOR THE FINE MESH, REPEATING SECTION

Stresses	Load Condition No. 1	Load Conditions 1 and 2	Load Condition No. 3
σ_{xx} (psi)	-.473741E+03	-.595979E+03	-.101403E+06
σ_{yy} (psi)	-.840852E+03	.130359E+04	-.447373E+05
σ_{zz} (psi)	-.922094E+03	-.126150E+04	-.796674E+04
σ_{xz} (psi)	.313721E+03	.102645E+04	.151684E+05

F. THERMAL GRADIENTS.

The previous change in the model made the thermal gradient in the area of the weld very flat. It was desired to determine what effect different thermal gradients on the right

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ORIGINAL  D.262

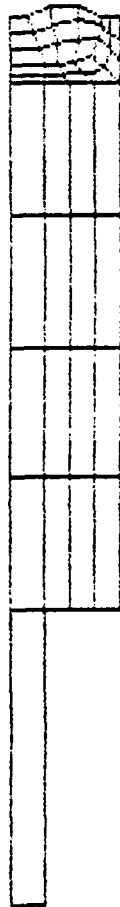
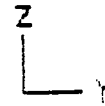


Figure 25. Coarse mesh, repeating section

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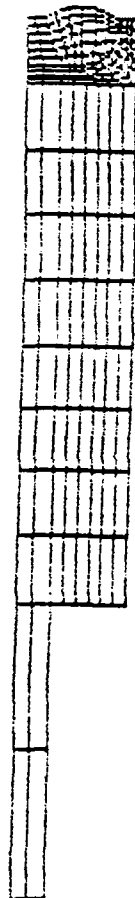
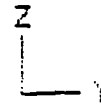



Figure 26. Fine mesh, repeating section

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ORIGINAL  D.262

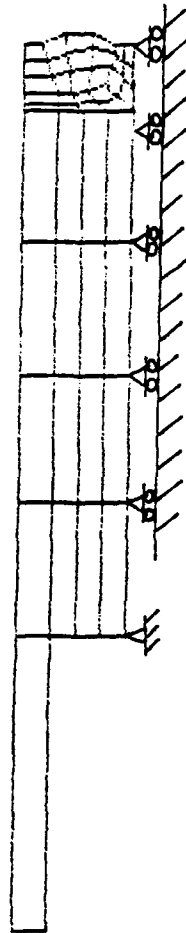
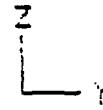


Figure 27. Rolling boundary conditions

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DEFORMED  D.0000847

TIME 2.000

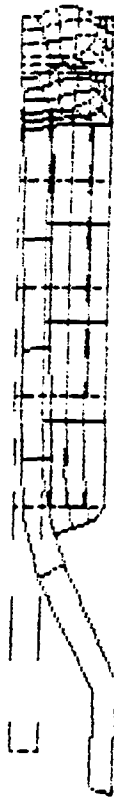
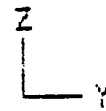



Figure 28. Original and deformed shape of coarse mesh for load condition no.1, repeating section

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DEFORMED  D.000250

TIME 2.000

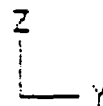


Figure 29. Original and deformed shape of coarse mesh for load condition no. 2 ,
repeating section

REPEAT HEADER

ORIGINAL L D.341
DEFORMED L D.00559
TIME 2.000

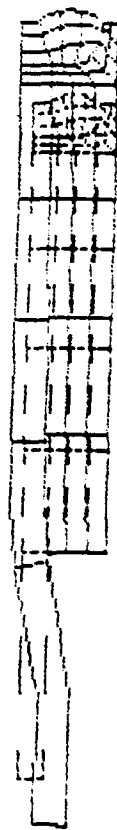
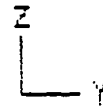


Figure 30. Original and deformed shape of coarse mesh for load condition no. 3 ,
repeating section

hand side of the model would have on the results. The maximum stresses for a thermal gradient of 100, 10, 2 and 0 degrees F are shown in Table 9.

Table 9. MAXIMUM STRESSES FOR DIFFERENT THERMAL GRADIENTS

Thermal Gradient	σ_{xx} (psi)	σ_{yy} (psi)	σ_{zz} (psi)	σ_{yz} (psi)
100°F	-.997953E+05	-.373304E+05	-.454326E+04	.104692E+05
10°F	-.998908E+05	-.373122E+05	-.454017E+04	.104358E+05
2°F	-.998993E+05	-.373106E+05	-.453990E+04	.104328E+05
0°F	-.999015E+05	-.373102E+05	-.453983E+04	.104321E+05

The results were surprising. The size of the gradient had very little effect on the magnitude of the stresses. The stresses remained very high even for a zero thermal gradient. A study of these results indicated the need for different boundary conditions. Restricting the right hand side of the model to move in the z direction only was not allowing the header to expand outwardly in the radial direction. The header was being forced to expand into the center of the tube which was creating the high stresses in the area of the weld.

The boundary condition chosen is shown in Figure 31 on page 45. The bottom left hand node of the header at the boundary of the tube and the header was fixed. The rest of the bottom nodes of the header were allowed to roll in the y direction. The contact surface option of ADINA could not be used for this solution. The nodes on the boundary were constrained together in the y direction and allowed to move with respect to one another in the z direction, modeling a contact surface with zero coefficient of friction. This was with the exception of the bottom left hand node of the header.

An iterative procedure was then followed to determine the solution of the problem. After each run of the problem, the bottom left hand node of the header was given a permanent displacement to the right equal to the displacement of the corresponding node of the tube at the boundary. This was continued until no overlap occurred. The original and deformed shapes of the final iteration for load condition no. 3 are shown in Figure 32 on page 46. As can be seen, the model is now expanding uniformly in the

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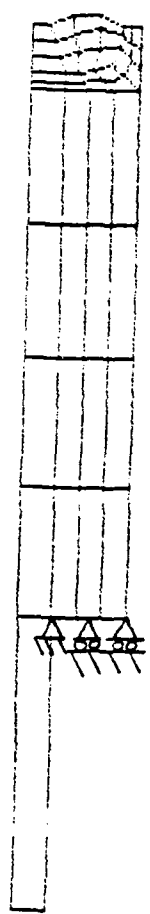
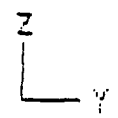




Figure 31. Bottom rolling boundary condition

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DEFORMED  D. 00369
TIME 2.000

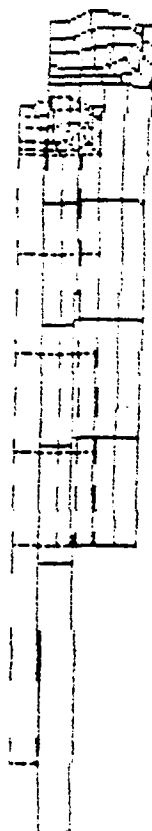
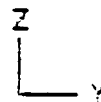


Figure 32. Original and deformed shape of the fine mesh for load condition no. 3, expanding radially and axially

axial direction and outwardly in the radial direction. The maximum stresses in the weld area of the coarse mesh for load condition no. 3 are shown in Table 10.

Table 10. MAXIMUM STRESSES FOR THE COARSE MESH, EXPANDING RADially

Stresses	Load Condition No. 3
σ_{xx} (psi)	.460123E+03
σ_{yy} (psi)	-.932915E+02
σ_{zz} (psi)	.102227E+03
σ_{xy} (psi)	-.584215E+02

These results are much lower than those obtained for the previous model and it is believed that these results are closer to the actual stresses. However, it is also believed that this model is not a true repeating section in that the expansion of the neighboring tubes and the rest of the header will have an effect on this section, probably raising the stresses in the weld area. The need for a 3-dimensional model was indicated.

A 3-dimensional model was also desired in order to apply non-symmetric loading to the model. The major difference between this header and other headers with a similar cross section is that this header is not sectionalized. It has one inlet/outlet header and one intermediate header. This results in a larger temperature gradient from the inlet to the outlet side. A 3-dimensional model would allow this type of non-symmetric loading to be studied.

V. 3-DIMENSIONAL SOLID FINITE ELEMENT MODEL

A. MODEL DEVELOPMENT

The repeating section chosen to be modeled was one tube row. Only half of the row was modeled due to symmetry. The welds were assumed to have the same material properties as the tube and header and no attempt was made to substructure the weld area.

The element chosen to model the structure was the 20-node isoparametric solid element. This element is shown in Figure 33 on page 49.

Figure 34 on page 50 shows the modeling of one tube. Each tube consists of sixty elements. Figure 35 on page 51 shows how the header was connected to the tube. A contact surface was placed between the tube and the header. The entire model is shown in Figure 36 on page 52. Figure 37 on page 53 is another view of the model showing the location of the origin of the axis. The model consists of 288 elements and 1957 nodes. Figure 38 on page 54, Figure 39 on page 55 and Figure 40 on page 56 show a view of the model looking from the x, y and z directions.

The hidden line computer code developed by David R. Hedgley, Jr. of the Ames Research Center was utilized to check the model. Different views are shown in Figure 41 on page 57 and Figure 42 on page 58.

The following boundary conditions were imposed. Every node in the $y=0$ plane was constrained to roll in that plane. In addition the node at the origin was fixed and the nodes above the origin along the positive x axis were constrained to roll only in the x direction.

B. INTERNAL PRESSURE LOAD

The internal pressure load of 700 psi was applied to the inside nodes of the tube and header. It was then discovered that ADINA does not optimize the band-width of contact surfaces. The bandwidth of this model is approximately 2700 and the problem was too large to be solved on the VAX/VMS operating system. The ADINA-IN input file for the contact surfaces is listed in Appendix II for reference.

A contact surface was then modeled by constraining the corresponding nodes of the tube and header to move together in the y and z direction, but allowing them to move relative to one another in the x direction. This assumes a coefficient of friction of zero and reduced the bandwidth to approximately 1000.

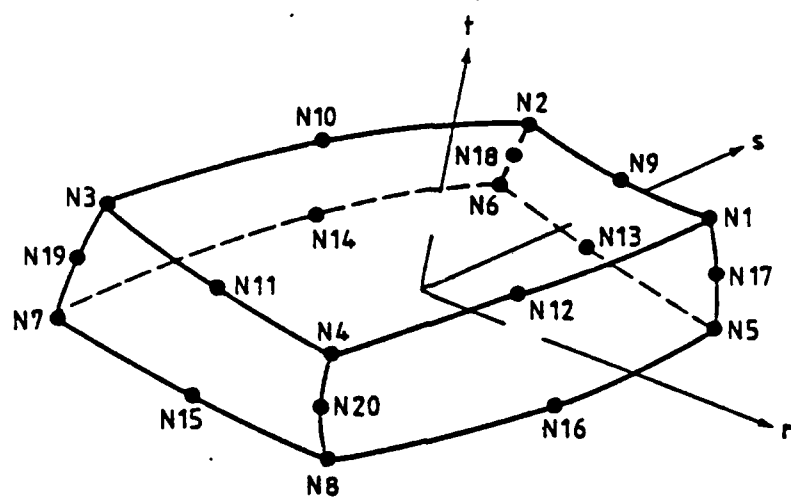



Figure 33. 20-node isoparametric solid element

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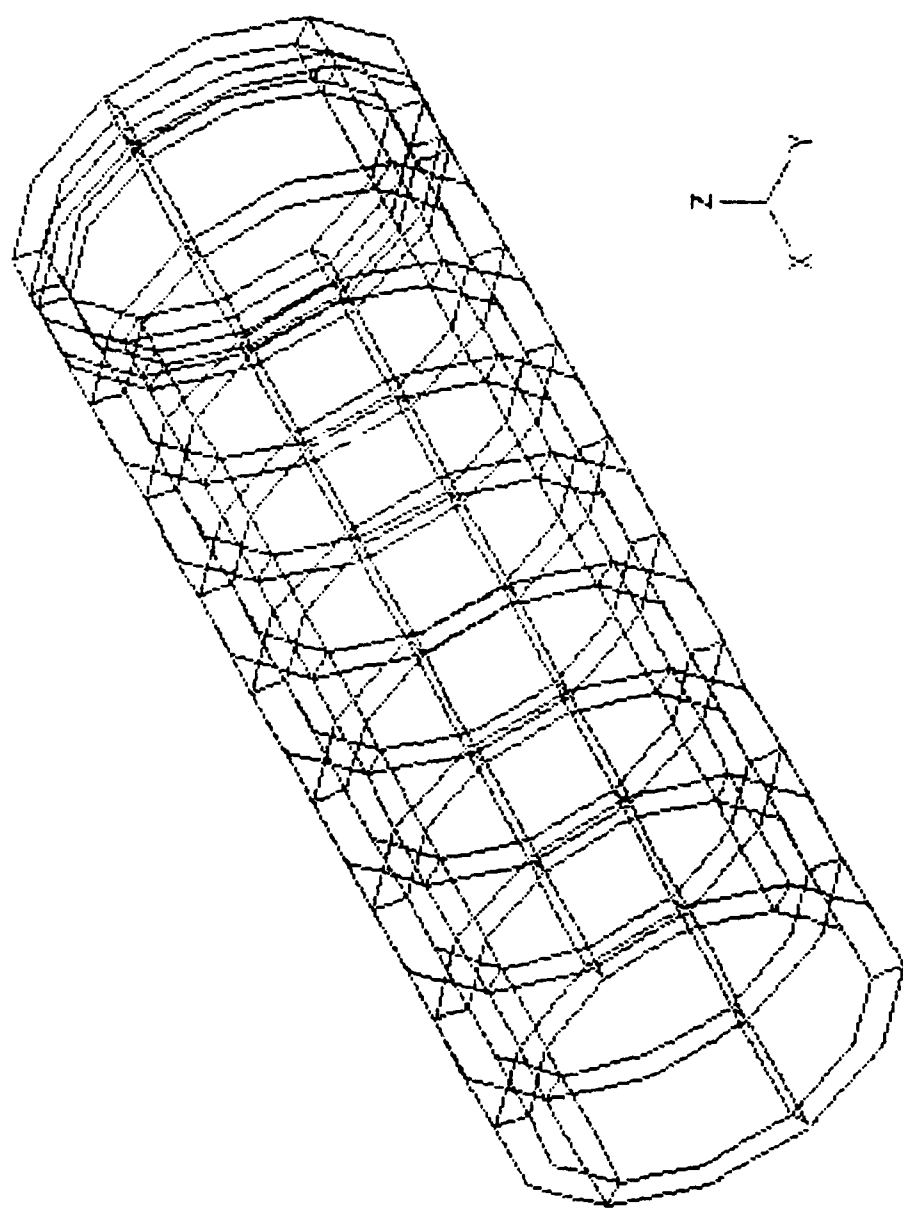



Figure 34. Mesh for one tube

3D-HEADER
ORIGINAL  D.194

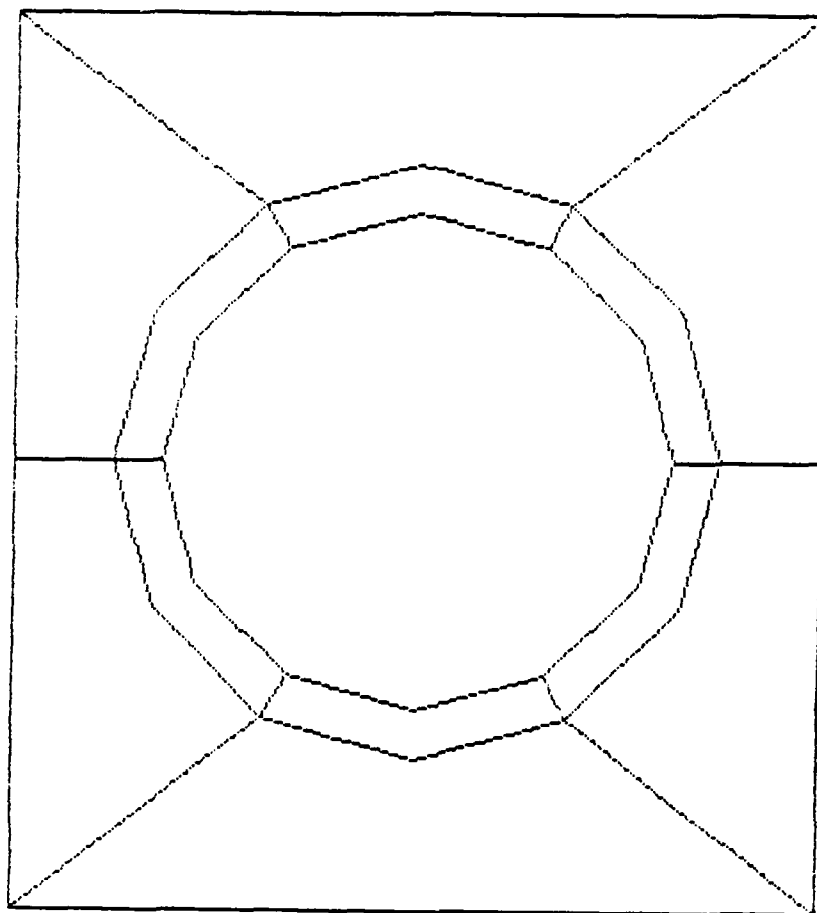
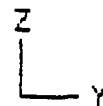


Figure 35. Attachment of header to a tube

3D-HEADER

ORIGINAL  D.715

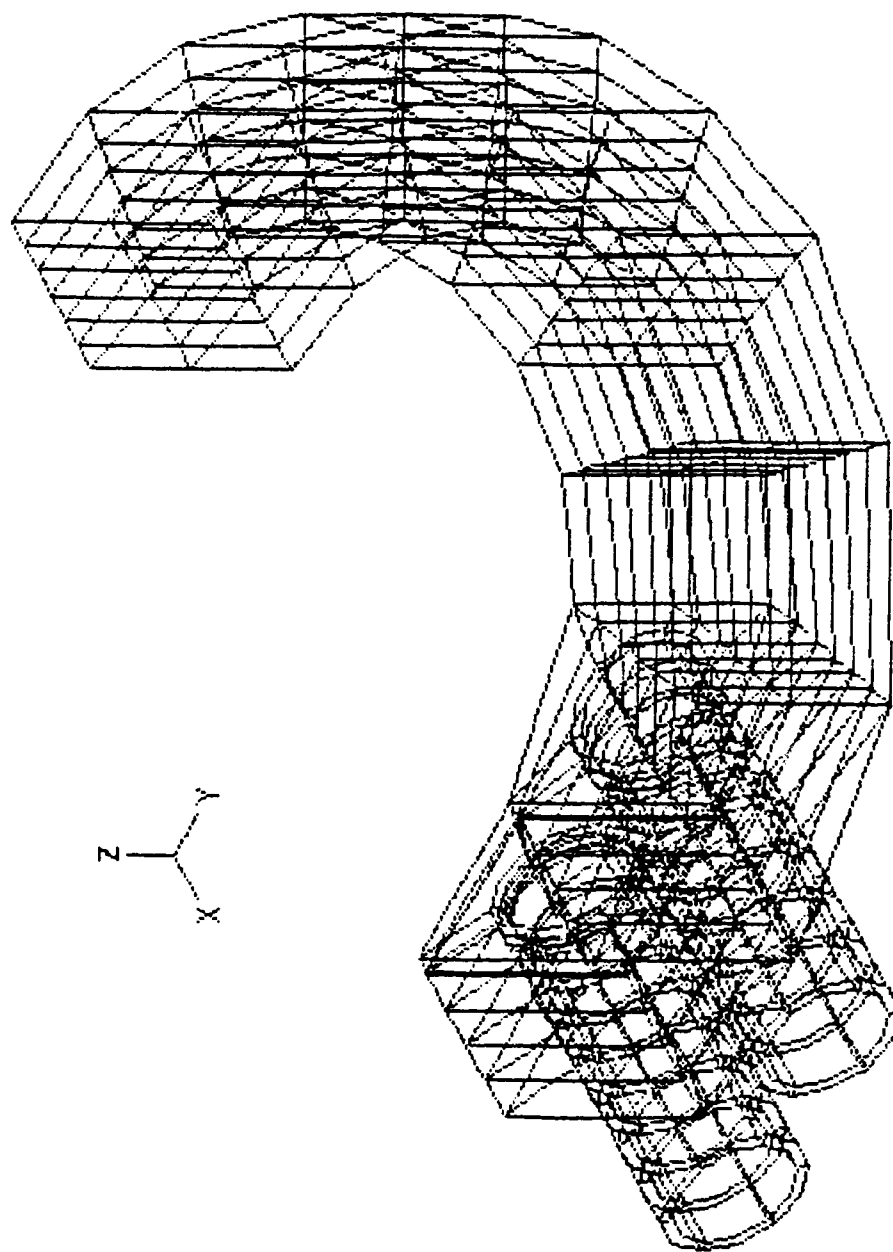
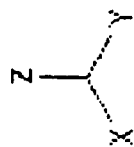


Figure 36. 3-dimensional model

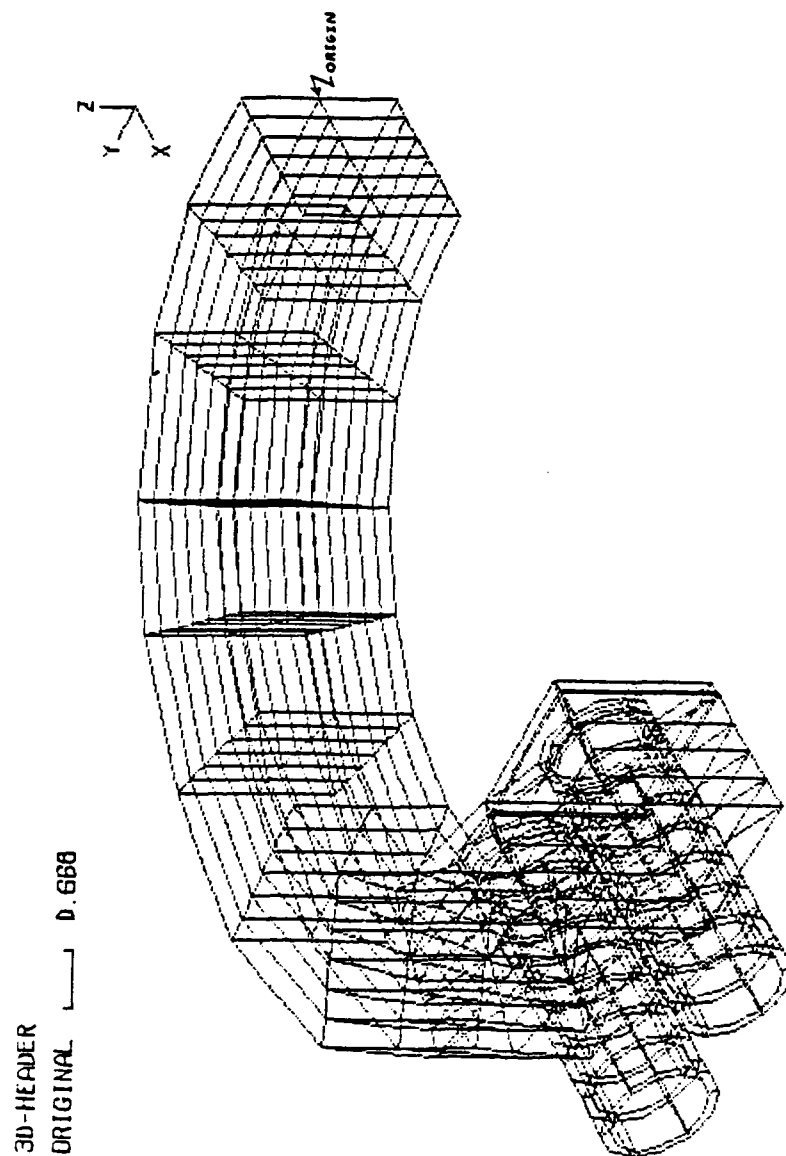


Figure 37. 3-dimensional model showing the location of the origin

3D-HEADER
ORIGINAL D. 956

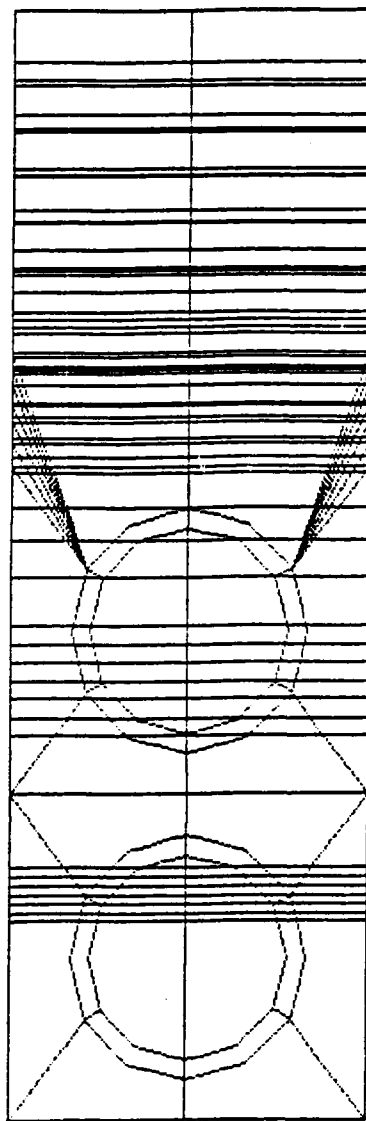


Figure 38. View of 3-dimensional model from the x direction

3D-HEADER
ORIGINAL  0.709

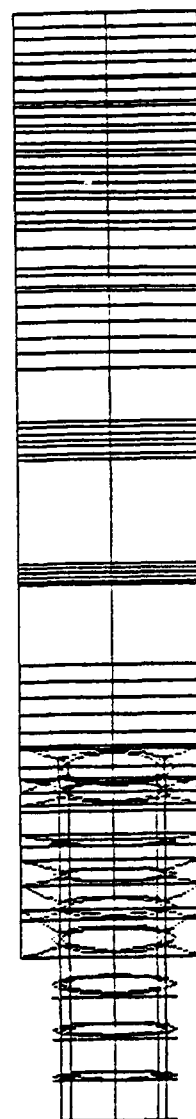


Figure 39. View of 3-dimensional model from the y direction

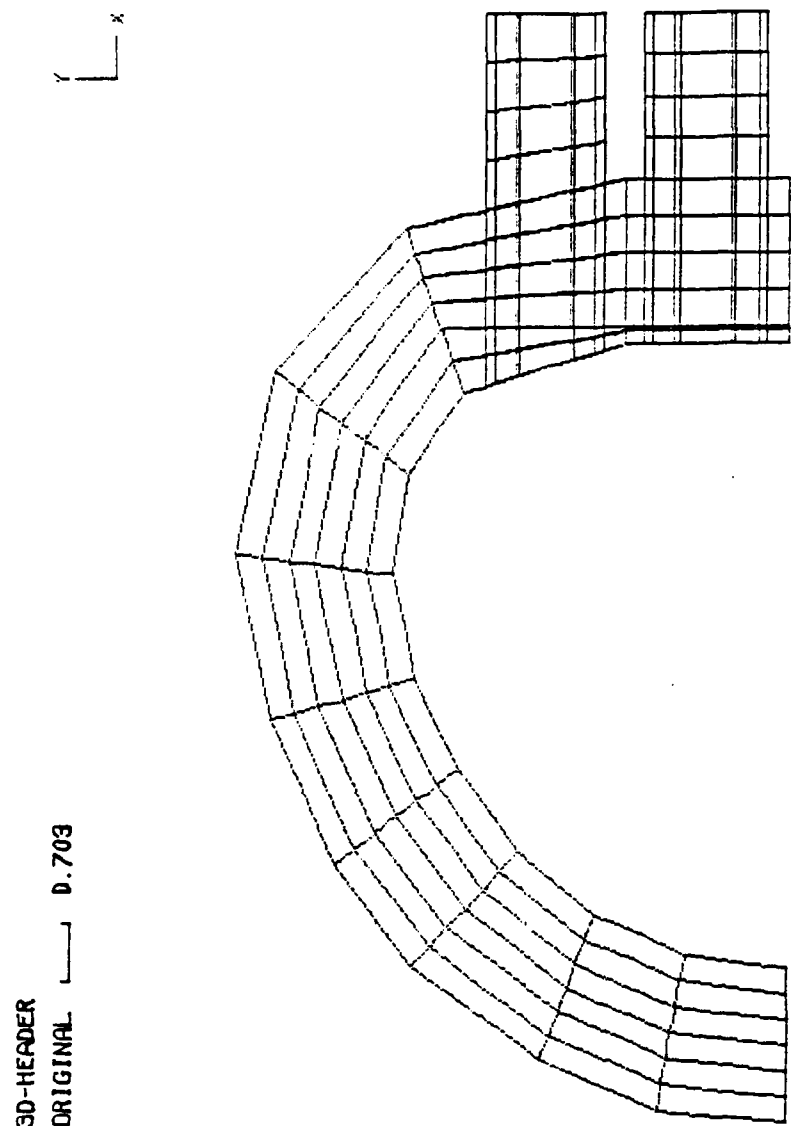


Figure 40. View of 3-dimensional model from the z direction

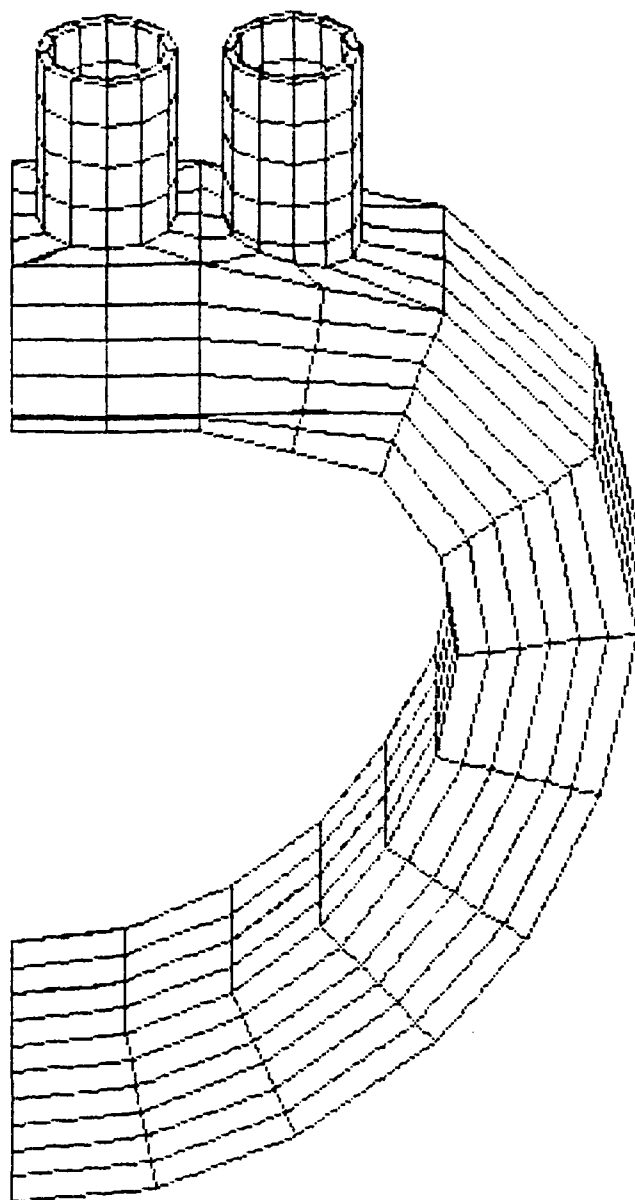


Figure 41. View of model using hidden line removal

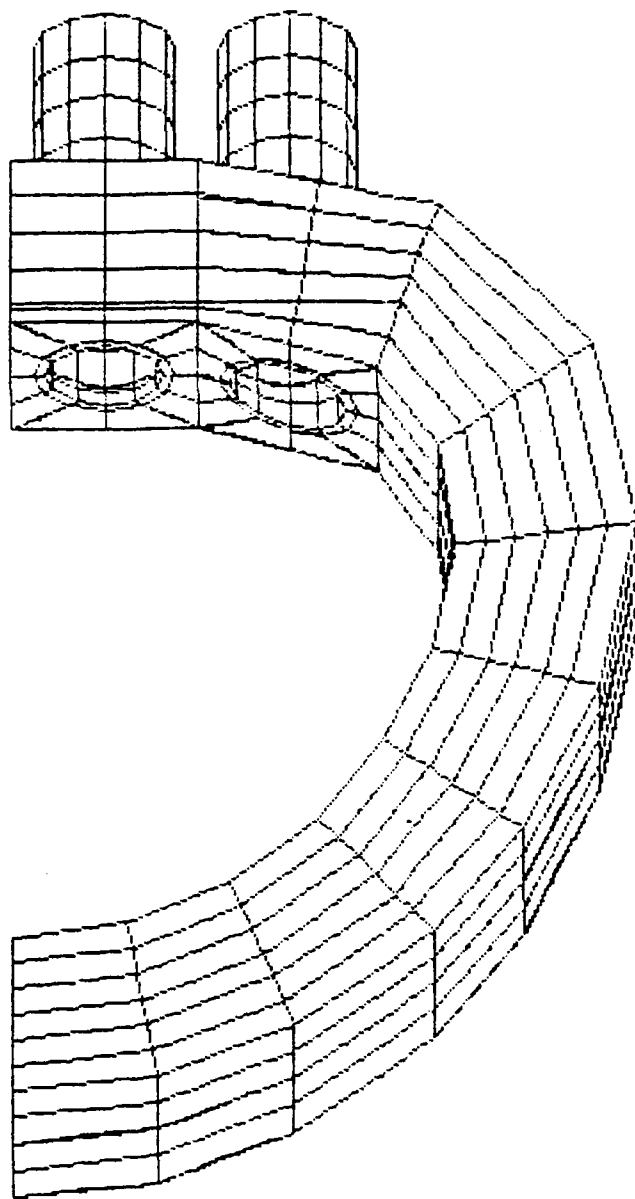


Figure 42. Another view of the model using hidden line removal

It then became necessary to enforce symmetry on either side of the $z=0$ plane. Every node in this plane was constrained to roll in the plane. It is understood that this enforced symmetry is more costly than modeling only half of the structure. The entire section was modeled to allow a study of unsymmetric temperature loading. The ADINA-IN input file is listed in Appendix I.

Figure 43 on page 60 shows the original and deformed shapes for this loading condition. Table 11 lists the maximum stresses in the area of the weld.

Table 11. MAXIMUM STRESSES
3-DIMENSIONAL MODEL

Stresses	Internal pressure load
σ_{xx} (psi)	-.984072E + 03
σ_{yy} (psi)	-.111941E + 04
σ_{zz} (psi)	.184639E + 04
σ_{xy} (psi)	-.487366E + 03
σ_{yz} (psi)	.235319E + 03
σ_{xz} (psi)	-.814224E + 03

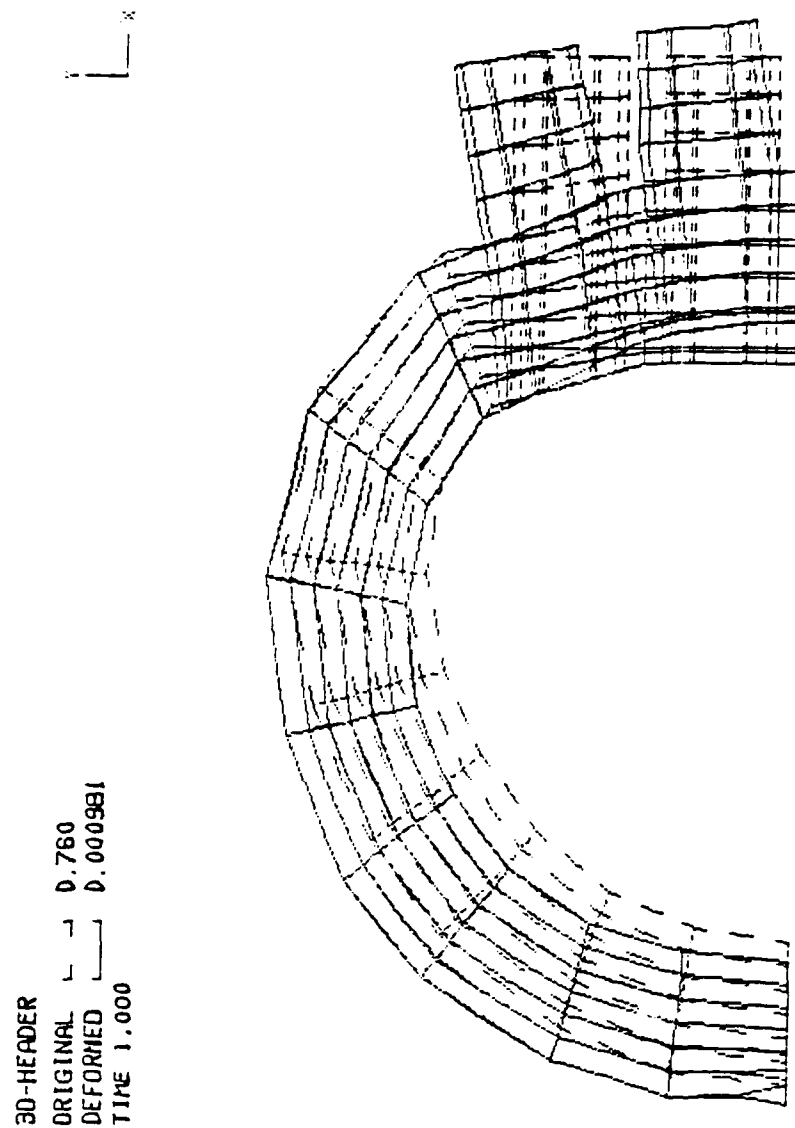


Figure 43. Original and deformed shape of the 3-dimensional model, internal pressure load

VI. CONCLUSIONS

A. DISCUSSION OF RESULTS

The maximum stresses obtained for the axisymmetric model were between 137 (ksi) and .46 (ksi), dependent upon the boundary conditions imposed. The axisymmetric model does not adequately represent the physical boundary conditions and the actual stress is somewhere between these two figures.

A 3-dimensional model has been developed which allows an accurate representation of the boundary conditions. The maximum stress for the internal pressure load in the area of the weld was 1.85 (ksi). This model also allows non-symmetric loading to be applied.

B. OPPORTUNITIES FOR FURTHER RESEARCH

The next step in solving this problem is to obtain results for the thermal loading. The initial loading should be static but the transient loading should also be studied. In studying the transient problem, a consideration should be made of the temperature dependence of the material properties of the structure. Another area for research would be the consideration of creep.

The major attempt of this report was to develop a 3-dimensional model which could be used to investigate the failures that have occurred in the superheater header attachment welds. This model can also be used to investigate any needed design changes. It is hoped that this model will be used for further investigation of this problem.

APPENDIX A. ADINA-IN PROCEDURES

These procedures are for the VAX/VMS operating system utilizing the plot-10 graphics library.

To operate ADINA-IN in the batch mode, a data file which defines the problem must be created. Assign this data file to a logical unit number and run ADINA-IN.

Now create the database. All of the input to ADINA-IN is stored on this database. The Command is DATAB CREATE. Next read the data file by using the command: READ (logical unit number).

The model can now be plotted using the frame, mesh and plot commands as explained in [Ref. 2: pp. 5.25-1 to 5.25-20]. The CONTROL command must be set to the correct parameters to use graphics. For the VAX/VMS operating system utilizing the plot-10 graphics library, set baudrate=4800, system=2, device=2 and option=1024. Changes can be made to the model by exiting and changing the data file or by typing in changes interactively. All changes will be saved on the database.

When the problem is properly defined, create the ADINA input file using the command ADINA. Exit ADINA-IN by typing in END.

ADINA-IN will have created two files. FOR001.DAT is the database. This database can best be saved by renaming it to a unique file name. To make changes assign the database to logical unit FOR001 and run ADINA-IN. Use the command DATABASE OPEN and then make any necessary changes.

The second file created by ADINA-IN is FOR002.DAT and this is the input file for ADINA. This file should be renamed to a unique filename and assigned to logical unit number FOR005. Assign an output file to logical unit number FOR006. Ensure there are no FOR*.* files in the directory as this can interfere with the operation of ADINA. Now run ADINA. The requested results will be printed in the assigned output file.

APPENDIX B. ADINA-IN PROCEDURES FOR THE THERMAL STRESS PROBLEM

ADINA-IN is also used for creating the input file for ADINAT. This particular version of ADINA-IN will not create the entire file which necessitates some minor editing of the ADINAT input file before ADINAT can be run.

The temperature loading in this thesis was done by using the LOADS TEMPERATURE command in the ADINA-IN input file. After the problem is defined in ADINA-IN, use the ADINAT command to generate the ADINAT input file.

ADINA-IN again creates two files. FOR001.DAT is the database and FOR002.DAT is the ADINAT input file. Rename these files.

The ADINAT input file must now be edited. Search for the line which has the letters 'KKKKKK'. This line must be deleted and the specific heat entered. The line just above this line must also be deleted and the thermal conductivity entered. Exit the ADINAT input file.

Assign the ADINAT input file to logical unit number FOR005 and assign an output file to logical unit number FOR006. Run ADINAT. ADINAT creates several files, most of which are used for the operation of the code. The final nodal point temperatures and heat fluxes are printed in the output file. The nodal temperatures are also stored in FOR056.DAT. This file must be renamed to a unique filename.

ADINA-IN must now be run again to create the input file for ADINA to solve for the thermal stresses. Assign the nodal temperature file to logical unit number FOR056.

The ADINA-IN input file must be changed. Delete the LOADS TEMPERATURE command and indicate on the MASTER command line that ITP56=1. Run ADINA-IN and read the changed input file. Ensure all FOR*.* files are deleted before doing this. Then use the ADINA command to create the ADINA input file. ADINA can now be run as explained in the last paragraph of Appendix A.

APPENDIX C. ADINA-PLOT PROCEDURES

ADINA created several files besides the assigned output file. Most of these files are used for the operation of the code. File FOR060.DAT is the porthole file for ADINA-PLOT.

Before running ADINA-PLOT, ensure all FOR001.DAT files are deleted and that logical unit number FOR001 is deassigned. Run ADINA-PLOT. First create the database using the DATABASE CREATE command. All of the information in the porthole file is now loaded.

Ensure the control information is correct. The model can now be plotted and the results listed as explained in [Ref. 6].

Exit ADINA-PLOT with the command END. ADINA-PLOT will have created a database with filename FOR001.DAT. Rename this file. To look at the results again, assign the database to logical unit number FOR001 and use the DATABASE OPEN command when running ADINA-PLOT.

APPENDIX D. VERIFICATION OF ADINA - CANTILEVER BEAM PROBLEM

The cantilever beam problem modeled is shown in Figure 44 on page 66. A moment of 360 k-in is applied to the end of a cantilever beam which is 48 inches long and 12 inches wide.

The classical equations as listed in [Ref. 7: pp. 215 and 737] are:

$$\sigma = My/I \quad (A.1)$$

$$\delta_b = M_o L^2 / 2EI \quad (A.2)$$

The finite element model showing both the original and deformed shapes is shown in Figure 45 on page 67. A nine node plane stress element was used. Table 12 shows a comparison of the results for several nodes. As can be seen, ADINA yields exact results for this problem. The following is the ADINA-IN input file:

Table 12. COMPARISON OF RESULTS

Comparison	Beam Theory	ADINA
displacement of node 10 (inch)	.096	.096
stress at node 1 (psi)	15	15
stress at node 12 (psi)	-15	-15

```

DATAB CREATE
HEAD ' CANTILEVER BEAM'
*
MASTER REACTIONS=YES IDOF=100111
PRINTOUT MAX
ANALYSIS TYPE=STATIC
*
COORDINATES / ENTRIES NODE X Y Z
1 0 0 0
5 0 48 0
15 0 48 12
11 0 0 12

```

*

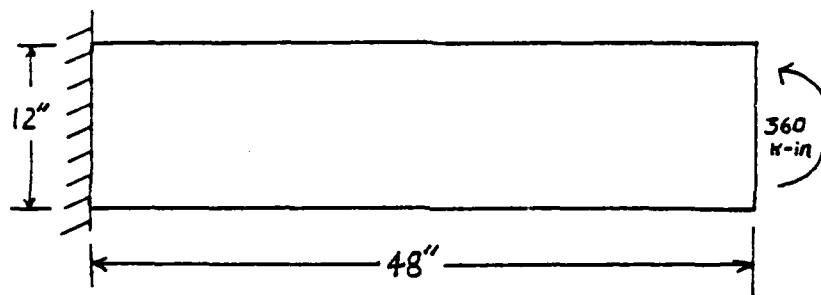


Figure 44. Cantilever beam with concentrated moment

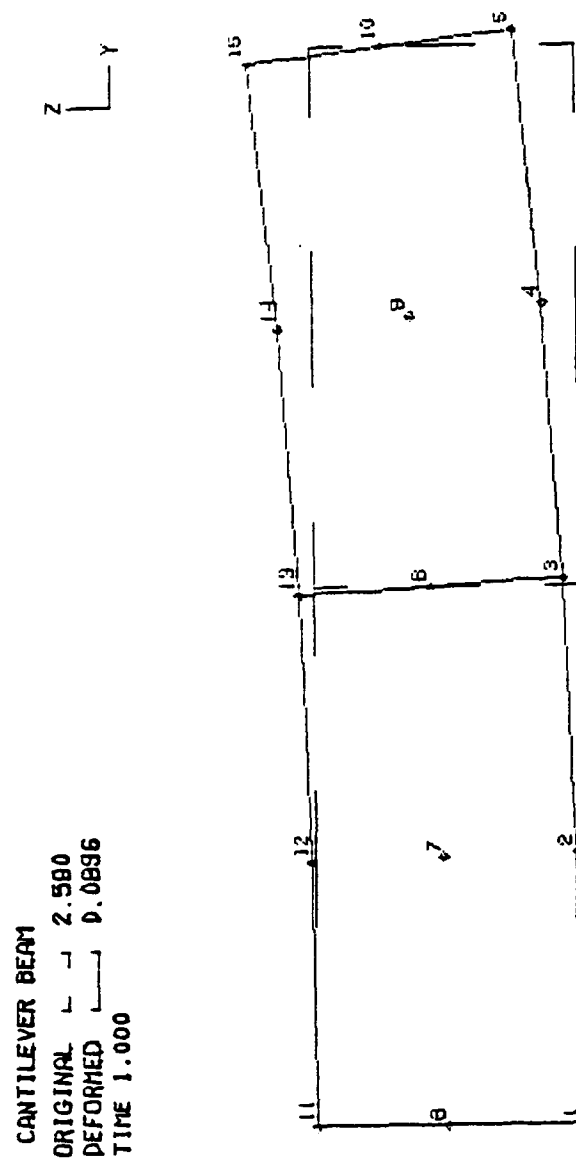


Figure 45. Original and deformed shapes for the cantilever beam

MATERIAL 1 ELASTIC E=3E4 NU=0.0 D=1
 *
 EGROUP 1 PLANE SUBTYPE=STRESS2 RESULTS=STRESSES INT=6
 STRESSTABLE N=1 P1=1 P2=2 P3=3 P4=4 P5=5 P6=6 P7=7 P8=8 P9=9
 EDATA / ENTRIES EL THICK
 1 1
 2 1
 GSURFACE 1 5 15 11 EL1=2 EL2=1 NODES=9
 *
 BOUNDARIES 111111 TYPE=NODES / 1 6 11
 *
 LOADS CONCENTRATED
 5 2 30
 15 2 -30
 *
 CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024
 CONTROL PLOTE=W PLOTS=N

APPENDIX E. VERIFICATION OF ADINA - ANNULAR DISK PROBLEM

The classical equations for the thermal stresses are listed in [Ref. 8 : p. 259]:

$$\sigma_r = \alpha E \left[-1/r^2 \int_a^r T r dr + (r^2 - a^2)/r^2 (b^2 - a^2) \int_a^b T r dr \right] \quad (E.1)$$

$$\sigma_\theta = \alpha E \left[-T + 1/r^2 \int_a^r T r dr + (r^2 + a^2)/r^2 (b^2 - a^2) \int_a^b T r dr \right] \quad (E.2)$$

where:

- a = inner radius, .63 inch
- b = outer radius, 3.0 inch
- T = temperature distribution, $399.569212 - 86.3436r + 12.16239867r^2$
- E = 29600000 psi
- $\alpha = .0000065 \text{ in/in/}^\circ\text{F}$

The finite element model showing the original and deformed shapes is shown in Figure 46 on page 70. A nine node plane axisymmetric element was used. Table 13 shows a comparison of the results. The comparison is very good. Most of the difference can be attributed to the fact that the results for the disk theory were computed at the nodes where as the results for ADINA are calculated at the Gaussian integration points. The following is the ADINA-IN input file:

Table 13. COMPARISON OF RESULTS

r (inch)	Stresses from Disk Theory (psi)	Stresses from ADINA (psi)
.63	$\sigma_\theta = -.139\text{E}+05$	$\sigma_\theta = -.120961\text{E}+05$
1.815	$\sigma_r = -.2577\text{E}+04$	$\sigma_r = -.253146\text{E}+04$
1.815	$\sigma_\theta = .1715\text{E}+04$	$\sigma_\theta = .169057\text{E}+04$
3.0	$\sigma_\theta = .534\text{E}+04$	$\sigma_\theta = .511644\text{E}+04$

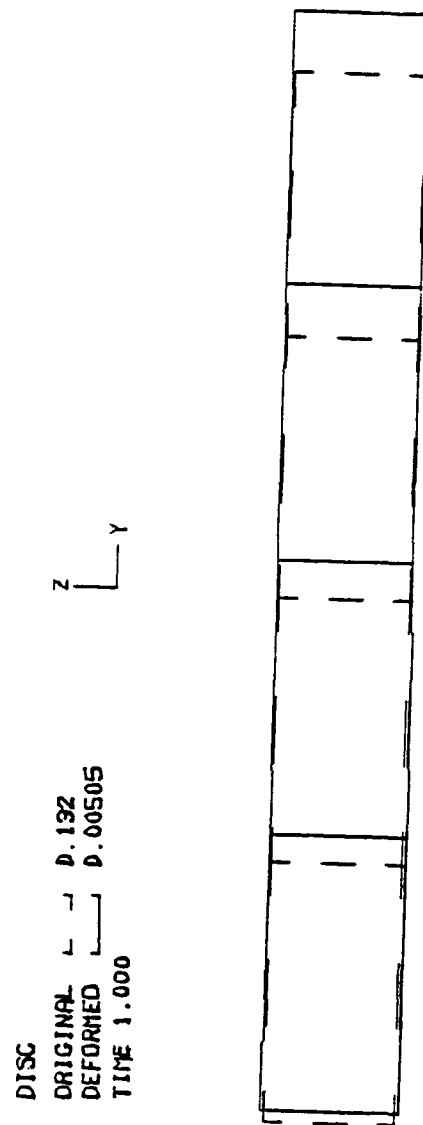


Figure 46. Original and deformed shapes for the disk

DATAB CREATE
HEAD 'DISC'
CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024
CONTROL PLOTE=WAIT PLOTS=NO

★
COORDINATES / ENTRIES

NODE	X	Y	Z
1	0	.63	0
2	0	.63	.3
3	0	3.0	0
4	0	3.0	.3

★
MATERIAL 1 THERMO-ELASTIC
-2000 29600000 .3 .0000065
2000 29600000 .3 .0000065

★
EGROUP 1 PLANE SU=AXISYMM
GSURFACE 4 2 1 3 EL1=4 EL2=1 NO=9

★
MASTER REACTIONS=YES IDOF=100111
PRINTOUT MAX
ANALYSIS TYPE=STATIC

★
BOUNDARIES 101111 TYPE =NODES/12 STEP 1 TO 20

★
LOADS TEMP TR=0

1	350
2	350
20	350
3	250
4	250
12	250

★

APPENDIX F. VERIFICATION OF ADINA - CHIMNEY PROBLEM

The chimney problem modeled is shown in Figure 47 on page 73. The inside surface is at a uniform temperature of 650 degrees C and the outside surface is at a temperature of 150 degrees C.

The finite element model is shown in Figure 48 on page 74. A nine node plane stress element was used.

A finite difference temperature distribution of this problem was obtained from Professor G. Cantin of the Naval Postgraduate School. The results had converged to the exact solution. Table 14 shows a comparison of the results for several nodes. The comparison is very good. The following is the ADINA-IN input file:

Table 14. RESULTS FOR CHIMNEY PROBLEM

Node	Temperatures from Finite Difference (°C)	Temperatures from ADINA (°C)
10	3.6933026E+02	3.68497E+02
12	3.4661577E+02	3.39891E+02
15	2.5149076E+02	2.52863E+02
16	3.4891941E+02	3.42042E+02
17	3.8732346E+02	4.01579E+02

```
DATAB CREATE
HEAD 'CHIMNEY'
CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024
CONTROL PLOTE=WAIT PLOTS=NO
```

```
*
COORDINATES/ENTRIES NODE X Y Z
1 0 2.0 1.5
2 0 1.0 1.5
3 0 1.0 1.0
4 0 2.0 1.0
5 0 0 1.0
6 0 0 0
7 0 2.0 0
```

```
*
MATERIAL 1 THERMO-ELASTIC
-2000 29600 .3 .0000065
2000 29600 .3 .0000065
```

```
*
```

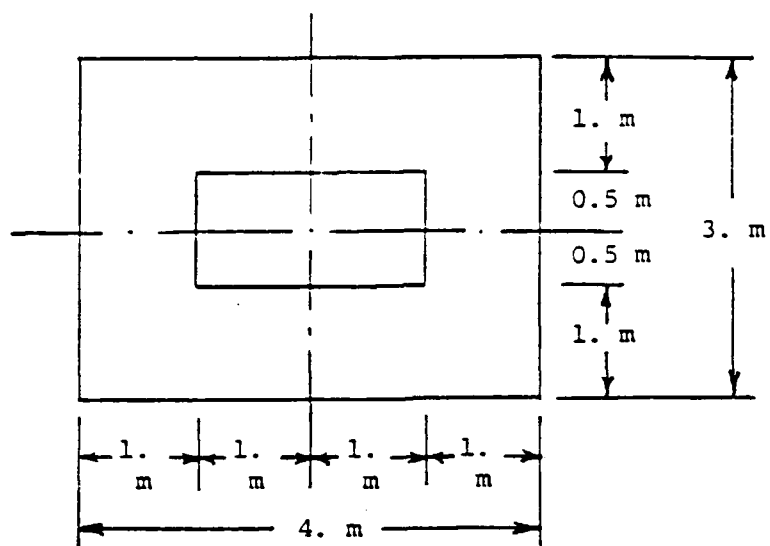


Figure 47. Chimney problem

CHIMNEY
ORIGINAL — D. 129

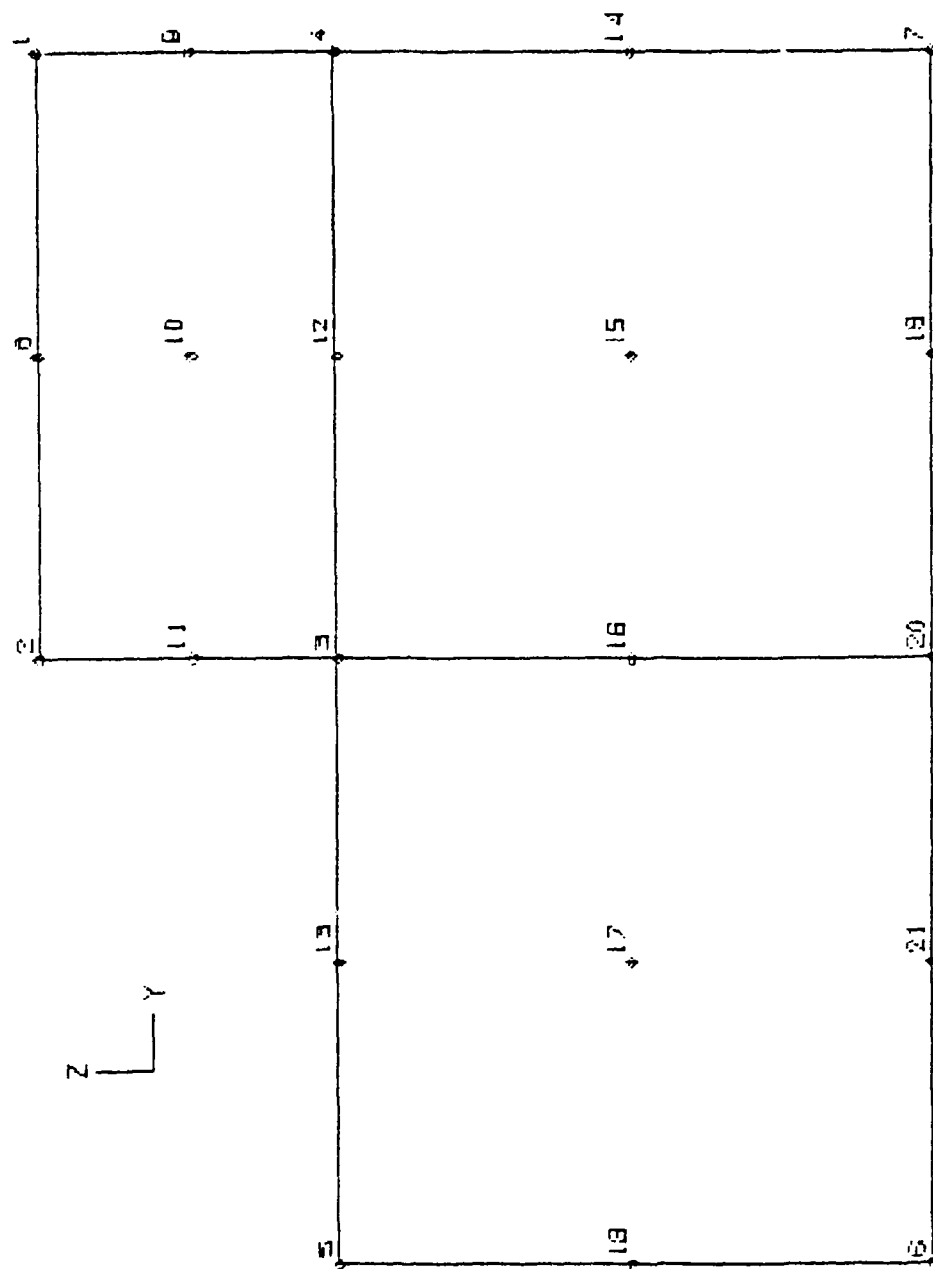


Figure 48. Finite element model of the chimney

```

EGROUP 1 PLANE SU=STRESS2
EDATA
1 1
2 1
3 1
GSURFACE 1 2 3 4 EL1=1 EL2=1 NO=9
GSURFACE 4 5 6 7 EL1=2 EL2=1 NO=9
*
MASTER REACTIONS=YES IDOF=100111
PRINTOUT MAX
ANALYSIS TYPE=STATIC
*
LOADS TEMPERATURE
1 150
9 150
4 150
14 150
7 150
19 150
20 150
21 150
2 650
11 650
3 650
13 650
5 650
*

```

APPENDIX G. ADINA-IN INPUT FILE FOR THE AXISYMMETRIC PROBLEM

DATAB CREATE

*INPUT FOR THE COARSE MESH, 2-DIMENSIONAL AXISYMMETRIC MODEL

HEAD 'REPEAT HEADER'

CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024

CONTROL PLOTE=WAIT PLOTS=NO

*COORDINATES ARE DEFINED

COORDINATES / ENTRIES	NODE	X	Y	Z
	1	0	.63	3.0
	3	0	.693	3.0
	37	0	.63	2.844
	39	0	.75	2.844
	60	0	.63	2.782
	61	0	.75	2.782
	70	0	.63	1.0
	71	0	.75	1.0
	72	0	.75001	1.0
	85	0	.63	0.0
	87	0	.75	0.0
	100	0	.938	3.0
	200	0	.82	2.837
	152	0	.938	2.97333333
	204	0	.9322366	2.93059481
	211	0	.901	2.88
	210	0	.84513864	2.84814711
	207	0	.792	2.844
	209	0	.813	2.969
	126	0	.938	2.98666666
	178	0	.938	2.96
	208	0	.813	2.844
	206	0	.771	2.844
	50	0	1.0	3.0
	6	0	1.0	2.782
	75	0	1.0	1.0

*MATERIAL PROPERTIES ARE DEFINED

MATERIAL 1 THERMO-ELASTIC

-2000 29600000 .3 .0000065

2000 29600000 .3 .0000065

*THE ELEMENT IS DEFINED

EGROUP 1 PLANE SU=AXISYMM

*THE MESH FOR THE TUBE IS GENERATED

GSURFACE 3 1 37 39 EL1=1 EL2=3 NO=9 NC=B

GSURFACE 39 37 60 61 EL1=1 EL2=2 NO=9 NC=B

GSURFACE 61 60 70 71 EL1=1 EL2=4 NO=9 NC=B

GSURFACE 71 70 85 87 EL1=1 EL2=1 NO=9 NC=B

*NODES FOR THE WELD ARE GENERATED

LINE A 3 100 NCENTER=200 EL=3 M=1

LINE N 100 204 126 152 178

LINE A 204 211 NCENTER=209 EL=1 M=1

LINE C 100 211 204

LINE A 211 210 NCENTER=209 EL=1 M=1

LINE N 210 39 208 207 206

LINE C 211 39 210

*THE MESH FOR THE WELD IS GENERATED

GSURFACE 100 3 39 211 EL1=3 EL2=3 NO=9 NC=B

*NODES FOR THE HEADER ARE GENERATED

LINE C 100 39 204 211 210

LINE S 50 6 EL=3 M=1

LINE S 6 61 EL=3 M=1

LINE C 50 61 6

*THE MESH FOR THE HEADER IS GENERATED

GSURFACE 50 100 39 61 EL1=2 EL2=6 NO=9 NC=B
 GSURFACE 6 61 72 75 EL1=3 EL2=4 NO=9 NC=N
 *PORTIONS OF THE MODEL ARE DEFINED FOR PLOTTING
 EZONE TUBE01
 1 STEP 1 TO 3
 EZONE TUBE02
 4 STEP 1 TO 10
 EZONE WELD
 11 STEP 1 TO 19
 EZONE HEAD
 20 STEP 1 TO 43
 BZONE CLOSE XMIN=0 XMAX=0 YMIN=.63 YMAX=1.0 ZMIN=2.78 ZMAX=3.1
 *THE MASTER DEGREES OF FREEDOM ARE DEFINED
 MASTER REACTIONS=YES IDOF=100111 DT=1 NSTEP=2 ITP56=1
 PRINTOUT MAX
 ANALYSIS TYPE=STATIC
 *THE LOAD IS APPLIED IN 2 STEPS
 TIMEFUNCTION
 0 0
 2 1
 *
 TOL T=E I=15 DN=.00001
 *NODES ONLY USED FOR GENERATION ARE FIXED, THE BOTTOM RIGHT HAND
 *NODE OF THE HEADER IS FIXED
 BOUNDARIES 111111 TYPE=NODES / 200 209 75
 *THE NODES ON THE RIGHT HAND SIDE OF THE HEADER ARE CONSTRAINED TO
 *ROLL ONLY IN THE Z DIRECTION
 BOUNDARIES 110111 TYPE=NODES / 50 101 STEP 1 TO 105 6
 BOUNDARIES 110111 TYPE=NODES / 148 156 163 170 177 185 192
 *THE CONTACT SURFACE IS DEFINED
 CGROUP 1 CONTACT2 S=AXISYM
 CONTACTSURFACE 1
 61 155 162 169 176 184 191 198 72
 CONTACTSURFACE 2
 87 56 71 52 48 45 42 38 34 31 61
 CONTACTPAIR 1 T=2 C=1 F=0
 *THE INTERNAL PRESSURE LOADING
 LOADS ELEMENT
 1 -1 700
 STEP 1 TO
 10 -1 700
 1 2 700
 13 2 700
 12 2 700
 11 2 700
 21 2 700
 20 2 700
 *THE LONGITUDINAL TUBE LOAD
 LOADS CONCENTRATED
 85 3 -25.3
 59 3 -101.3
 87 3 -25.3
 *THE THERMAL LOAD
 LOADS TEMP TR=0
 50 350.0
 101 348.1833333
 102 346.3666667
 103 344.55
 104 342.7333333
 105 340.9166667

6 339.1
 148 327.9625
 156 316.825
 163 305.6875
 170 294.55
 177 283.4125
 185 272.275
 192 261.1375
 75 250
 1 350
 7 350
 10 350
 13 350
 16 350
 19 350
 37 350
 23 350
 26 350
 29 350
 60 350
 33 350
 36 350
 41 350
 44 350
 47 350
 51 350
 54 350
 70 350
 58 350
 85 350
 2 350
 3 350
 62 350
 63 350
 64 350
 65 350
 66 350
 100 350
 113 350
 112 350
 111 350

DATAB CREATE

*INPUT FOR THE FINE MESH, 2-DIMENSIONAL AXISYMMETRIC MODEL

HEAD 'HEADER'

CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024

CONTROL PLOTE=WAIT PLOTS=NO

*COORDINATES ARE DEFINED

COORDINATES / ENTRIES	NODE	X	Y	Z
	1	0	.63	3.0
	3	0	.693	3.0
	37	0	.63	2.844
	39	0	.75	2.844
	60	0	.63	2.782
	61	0	.75	2.782
	70	0	.63	1.0
	71	0	.75	1.0
	72	0	.75001	1.0
	85	0	.63	0.0
	87	0	.75	0.0
	100	0	.938	3.0
	200	0	.82	2.837
	152	0	.938	2.97333333
	256	0	.901	2.88
	248	0	.792	2.844

251	0	.813	2.969
126	0	.938	2.98666666
178	0	.938	2.96
250	0	.813	2.844
246	0	.771	2.844
50	0	1.0	3.0
6	0	1.0	2.782
75	0	1.0	1.0
113	0	.938	2.99333333
139	0	.938	2.98
165	0	.938	2.96666666
249	0	.8025	2.844
247	0	.7815	2.844
245	0	.7605	2.844

*MATERIAL PROPERTIES ARE DEFINED
 MATERIAL 1 ELASTIC E=296000000 NU=.3
 *THE ELEMENT IS DEFINED
 EGROUP 1 PLANE SU=AXISYMM
 *THE MESH FOR THE TUBE IS GENERATED
 GSURFACE 3 1 37 39 EL1=2 EL2=6 NO=9
 GSURFACE 39 37 60 61 EL1=2 EL2=4 NO=9
 GSURFACE 61 60 70 71 EL1=2 EL2=8 NO=9
 GSURFACE 71 70 85 87 EL1=2 EL2=2 NO=9
 *NODES FOR THE WELD ARE GENERATED
 LINE A 3 100 NCENTER=200 EL=6 M=1
 LINE N 100 178 113 126 139 152 165
 LINE A 178 256 NCENTER=251 EL=3 M=1
 LINE C 100 256 178
 LINE A 256 250 NCENTER=251 EL=3 M=1
 LINE N 250 39 249 248 247 246 245
 LINE C 256 39 250
 *THE MESH FOR THE WELD IS GENERATED
 GSURFACE 100 3 39 256 EL1=6 EL2=6 NO=9
 *NODES FOR THE HEADER ARE GENERATED
 LINE C 100 39 178 256 250
 LINE S 50 6 EL=6 M=1
 LINE S 6 61 EL=6 M=1
 LINE C 50 61 6
 *THE MESH FOR THE HEADER IS GENERATED
 GSURFACE 50 100 39 61 EL1=4 EL2=12 NO=9
 GSURFACE 6 61 72 75 EL1=6 EL2=8 NO=9 NC=N
 *PORTIONS OF THE TUBE ARE DEFINED FOR PLOTTING
 EZONE TUBE01
 1 STEP 1 TO 12
 EZONE TUBE02
 13 STEP 1 TO 44
 EZONE WELD
 45 STEP 1 TO 80
 EZONE HEAD
 81 STEP 1 TO 256
 BZONE CLOSE XMIN=0 XMAX=0 YMIN=.63 YMAX=1.094 ZMIN=2.688 ZMAX=3.1
 BZONE HEAD01 XMIN=0 XMAX=0 YMIN=1.094 YMAX=2.0 ZMIN=2.688 ZMAX=3.0
 BZONE HEAD02 XMIN=0 XMAX=0 YMIN=1.094 YMAX=2.0 ZMIN=1.0 ZMAX=2.688
 BZONE HEAD03 XMIN=0 XMAX=0 YMIN=.63 YMAX=1.094 ZMIN=1.0 ZMAX=2.688
 BZONE TUBE XMIN=0 XMAX=0 YMIN=.63 YMAX=.693 ZMIN=0 ZMAX=1.0
 *THE MASTER DEGREES OF FREEDOM ARE DEFINED
 MASTER REACTIONS=YES IDOF=100111 DT=1 NSTEP=2
 PRINTOUT MAX
 ANALYSIS TYPE=STATIC
 *THE LOAD IS APPLIED IN 2 STEPS
 TIMEFUNCTION
 0 0
 2 1
 *
 TOL T=E I=15 DN=.00001
 *NODES USED FOR GENERATION ARE FIXED, THE BOTTOM RIGHT HAND NODE
 *OF THE HEADER IS FIXED
 BOUNDARIES 111111 TYPE=NODES / 200 251 75

*THE NODES AT THE RIGHT HAND SIDE OF THE HEADER ARE CONSTRAINED TO
 *ROLL ONLY IN THE Z-DIRECTION
 BOUNDARIES 110111 TYPE=NODES/ 50 368 STEP 1 TO 378 6 558 STEP 13 TO 740
 *THE CONTACT SURFACES ARE DEFINED
 CGROUP 1 CONTACT2 S=AXISYM
 CONTACTSURFACE 1
 87 210 205 199 71 191 186 181 175 170 164 159 154 148 143 137 132 127 121 116 61
 CONTACTSURFACE 2
 61 570 583 596 609 622 635 648 661 674 687 700 713 726 739 752 72
 CONTACTPAIR 1 T=2 C=1 F=0
 *THE INTERNAL PRESSURE LOAD
 LOADS ELEMENT
 2 -1 700
 STEP 2 TO
 40 -1 700
 2 2 700
 1 2 700
 41 2 700
 STEP 1 TO
 46 2 700
 77 2 700
 STEP 1 TO
 80 2 700
 *THE LONGITUDINAL TUBE LOAD
 LOADS CONCENTRATED
 85 3 -12.66666666
 217 3 -50.66666666
 216 3 -25.33333333
 215 3 -50.66666666
 87 3 -12.66666666
 *THE THERMAL LOAD
 LOADS TEMP TR=0
 50 350
 368 349.0916667
 369 348.1833333
 370 347.275
 371 346.3666667
 372 345.4583333
 373 344.55
 374 343.6416667
 375 342.7333333
 376 341.825
 377 340.9166667
 378 340.0083333
 6 339.1
 558 333.53125
 571 327.9625
 584 322.39375
 597 316.825
 610 311.25625
 623 305.6875
 636 300.11875
 649 294.55
 662 288.98125
 675 283.4125
 688 277.84375
 701 272.275
 714 266.70625
 727 261.1375
 740 255.56875
 75 250
 1 350
 STEP 1 TO
 5 350
 218 350
 STEP 1 TO
 228 350

100 350
390 350
STEP 1 TO
396 350
50 350
11 350
STEP 5 TO
36 350
43 350
48 350
54 350
59 350
66 350
37 350
78 350
83 350
90 350
95 350
101 350
106 350
111 350
60 350
120 350
125 350
131 350
136 350
142 350
147 350
153 350
158 350
163 350
169 350
174 350
180 350
185 350
190 350
195 350
70 350
204 350
209 350
214 350
85 350
★

APPENDIX H. ADINA-IN INPUT FILE FOR THE 3-DIMENSIONAL CONTACT SURFACE

*THESE ARE THE CONTACT SURFACES FOR THE 3-DIMENSIONAL MODEL
CGROUP 1 CONTACT3

*CONTACTSURFACE 1 IS THE HEADER AROUND THE INNER TUBE
CONTACTSURFACE 1

1	123	124	970	969
2	969	970	982	981
3	981	982	994	993
4	993	994	69	68
5	124	125	971	970
6	970	971	983	982
7	982	983	995	994
8	994	995	70	69
9	125	126	1021	971
10	971	1021	1029	983
11	983	1029	1037	995
12	995	1037	71	70
13	126	127	1022	1021
14	1021	1022	1030	1029
15	1029	1030	1038	1037
16	1037	1038	72	71
17	127	128	1057	1022
18	1022	1057	1065	1030
19	1030	1065	1073	1038
20	1038	1073	73	72
21	128	129	1058	1057
22	1057	1058	1066	1065
23	1065	1066	1074	1073
24	1073	1074	74	73
25	129	130	1093	1058
26	1058	1093	1101	1066
27	1066	1101	1109	1074
28	1074	1109	75	74
29	130	131	1094	1093
30	1093	1094	1102	1101
31	1101	1102	1110	1109
32	1109	1110	76	75
33	131	132	1129	1094
34	1094	1129	1137	1102
35	1102	1137	1145	1110
36	1110	1145	77	76
37	132	11	1130	1129
38	1129	1130	1138	1137
39	1137	1138	1146	1145
40	1145	1146	6	77

41	11	122	1161	1130
42	1130	1161	1165	1138
43	1138	1165	1169	1146
44	1146	1169	67	6
45	122	123	969	1161
46	1161	969	981	1165
47	1165	981	993	1169
48	1169	993	68	67

*CONTACTSURFACE 2 IS THE HEADER AROUND THE OUTER TUBE
CONTACTSURFACE 2

1	585	586	1354	1323
2	1323	1354	1358	1331
3	1331	1358	1362	1339
4	1339	1362	531	530
5	586	587	1180	1354
6	1354	1180	1188	1358
7	1358	1188	1196	1362
8	1362	1196	532	531
9	587	588	1181	1180
10	1180	1181	1189	1188
11	1188	1189	1197	1196
12	1196	1197	533	532
13	588	589	1182	1181
14	1181	1182	1190	1189
15	1189	1190	1198	1197
16	1197	1198	534	533
17	589	590	1212	1182
18	1182	1212	1218	1190
19	1190	1218	1224	1198
20	1198	1224	535	534
21	590	591	1213	1212
22	1212	1213	1219	1218
23	1218	1219	1225	1224
24	1224	1225	536	535
25	591	592	1250	1213
26	1213	1250	1258	1219
27	1219	1258	1266	1225
28	1225	1266	537	536
29	592	593	1251	1250
30	1250	1251	1259	1258
31	1258	1259	1267	1266
32	1266	1267	538	537
33	593	594	1286	1251
34	1251	1286	1294	1259
35	1259	1294	1302	1267
36	1267	1302	539	538
37	594	473	1287	1286
38	1286	1287	1295	1294
39	1294	1295	1303	1302
40	1302	1303	468	539
41	473	584	1322	1287

42 1287 1322 1330 1295
 43 1295 1330 1338 1303
 44 1303 1338 529 468
 45 584 585 1323 1322
 46 1322 1323 1331 1330
 47 1330 1331 1339 1338
 48 1338 1339 530 529
 *CONTACTSURFACE 3 IS THE INNER TUBE
 CONTACTSURFACE 3
 1 195 237 238 196
 2 237 90 91 238
 3 90 297 298 91
 4 297 339 340 298
 5 339 381 382 340
 6 381 68 69 382
 7 196 238 239 197
 8 238 91 92 239
 9 91 298 299 92
 10 298 340 341 299
 11 340 382 383 341
 12 382 69 70 383
 13 197 239 240 198
 14 239 92 93 240
 15 92 299 300 93
 16 299 341 342 300
 17 341 383 384 342
 18 383 70 71 384
 19 198 240 241 199
 20 240 93 94 241
 21 93 300 301 94
 22 300 342 343 301
 23 342 384 385 343
 24 384 71 72 385
 25 199 241 242 200
 26 241 94 95 242
 27 94 301 302 95
 28 301 343 344 302
 29 343 385 386 344
 30 385 72 73 386
 31 200 242 243 201
 32 242 95 96 243
 33 95 302 303 96
 34 302 344 345 303
 35 344 386 387 345
 36 386 73 74 387
 37 201 243 244 202
 38 243 96 97 244
 39 96 303 304 97
 40 303 345 346 304
 41 345 387 388 346
 42 387 74 75 388
 43 202 244 245 203

44 244 97 98 245
 45 97 304 305 98
 46 304 346 347 305
 47 346 388 389 347
 48 388 75 76 389
 49 203 245 246 204
 50 245 98 99 246
 51 98 305 306 99
 52 305 347 348 306
 53 347 389 390 348
 54 389 76 77 390
 55 204 246 235 193
 56 246 99 8 235
 57 99 306 295 8
 58 306 348 337 295
 59 348 390 379 337
 60 390 77 6 379
 61 193 235 236 194
 62 235 8 89 236
 63 8 295 296 89
 64 295 337 338 296
 65 337 379 380 338
 66 379 6 67 380
 67 194 236 237 195
 68 236 89 90 237
 69 89 296 297 90
 70 296 338 339 297
 71 338 380 381 339
 72 380 67 68 381

*CONTACTSURFACE 4 IS THE OUTER TUBE

CONTACTSURFACE 4

1 657 699 700 658
 2 699 552 553 700
 3 552 759 760 553
 4 759 801 802 760
 5 801 843 844 802
 6 843 530 531 844
 7 658 700 701 659
 8 700 553 554 701
 9 553 760 761 554
 10 760 802 803 761
 11 802 844 845 803
 12 844 531 532 845
 13 659 701 702 660
 14 701 554 555 702
 15 554 761 762 555
 16 761 803 804 762
 17 803 845 846 804
 18 845 532 533 846
 19 660 702 703 661
 20 702 555 556 703
 21 555 762 763 556

22 762 804 805 763
23 804 846 847 805
24 846 533 534 847
25 661 703 704 662
26 703 556 557 704
27 556 763 764 557
28 763 805 806 764
29 805 847 848 806
30 847 534 535 848
31 662 704 705 663
32 704 557 558 705
33 557 764 765 558
34 764 806 807 765
35 806 848 849 807
36 848 535 536 849
37 663 705 706 664
38 705 558 559 706
39 558 765 766 559
40 765 807 808 766
41 807 849 850 808
42 849 536 537 850
43 664 706 707 665
44 706 559 560 707
45 559 766 767 560
46 766 808 809 767
47 808 850 851 809
48 850 537 538 851
49 665 707 708 666
50 707 560 561 708
51 560 767 768 561
52 767 809 810 768
53 809 851 852 810
54 851 538 539 852
55 666 708 697 655
56 708 561 470 697
57 561 768 757 470
58 768 810 799 757
59 810 852 841 799
60 852 539 468 841
61 655 697 698 656
62 697 470 551 698
63 470 757 758 551
64 757 799 800 758
65 799 841 842 800
66 841 468 529 842
67 656 698 699 657
68 698 551 552 699
69 551 758 759 552
70 758 800 801 759
71 800 842 843 801
72 842 529 530 843

APPENDIX I. ADINA-IN INPUT FILE FOR THE 3-DIMENSIONAL PROBLEM

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DATAB CREATE
HEAD '3D-HEADER'
CONTROL B=4800 SYSTEM=2 DEVICE=2 OPTION=1024
CONTROL PLOTE=WAIT PLOTS=NO
*THE COORDINATES FOR THE INNER TUBE ARE DEFINED
SYSTEM 1 TYPE=CYLINDRICAL X=9.4 Y=1 Z=1.125
COORDINATES /ENTRIES NODE R THETA XL
      1 .63 0 0
      2 .75 0 0
      3 .63 0 .156
      4 .75 0 .156
      5 .63 0 .218
      6 .75 0 .218
      7 .63 0 2.0
      8 .75 0 2.0
      9 .63 0 4.0
     10 .75 0 4.0
     11 .75001 0 2.0
LINE CYLINDRICAL 1 1 6 1
LINE CYLINDRICAL 2 2 6 1
LINE CYLINDRICAL 3 3 6 1
LINE CYLINDRICAL 4 4 6 1
LINE CYLINDRICAL 5 5 6 1
LINE CYLINDRICAL 6 6 6 1
LINE CYLINDRICAL 7 7 6 1
LINE CYLINDRICAL 8 8 6 1
LINE CYLINDRICAL 9 9 6 1
LINE CYLINDRICAL 10 10 6 1
LINE CYLINDRICAL 11 11 6 1
*THE MASTER DEGREES OF FREEDOM ARE DEFINED
MASTER REACTIONS=YES IDOF=000111
PRINTOUT V=MIN IO=0
ANALYSIS TYPE=STATIC
VIEW 1 X=1
VIEW 2 X=-1
VIEW 3 Y=1
VIEW 4 Y=-1
VIEW 5 Z=1
VIEW 6 Z=-1
*THE MATERIAL PROPERTIES ARE DEFINED
MATERIAL 1 ELASTIC E=29600000 NU=.3
*THE ELEMENT IS DEFINED
EGROUP 1 SOLID
STRESSTABLE 1 1 2 3 4 5 6 7 8 21
EDATA/ENTRIES EL TABLE PRINT
      1 1 YES
      STEP 1 TO
     288 1 YES
*THE MESH FOR THE INNER TUBE IS GENERATED
GVOLUME 10 10 9 9 8 8 7 7 EL1=6 EL2=1 EL3=4 NO=20
GVOLUME 8 8 7 7 6 6 5 5 EL1=6 EL2=1 EL3=4 NO=20
GVOLUME 6 6 5 5 4 4 3 3 EL1=6 EL2=1 EL3=1 NO=20
GVOLUME 4 4 3 3 2 2 1 1 EL1=6 EL2=1 EL3=1 NO=20
*THE COORDINATES FOR THE OUTER TUBE ARE DEFINED
SYSTEM 2 TYPE=CYLINDRICAL X=9.4 Y=3.0 Z=1.125

```

NO 4195 448

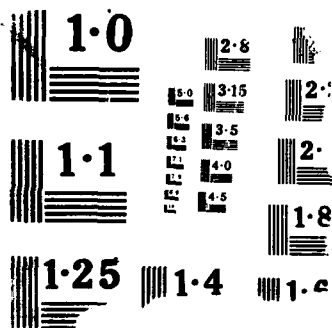
2-DIMENSIONAL AXISYMMETRIC AND 3- DIMENSIONAL FINITE
ELEMENT STRESS ANALYSIS OF THE LHA-1 CLASS SUPERHEATER
HEADER(U) NAVAL POSTGRADUATE SCHOOL MONTEREY CA
D R KITCHIN MAR 88

UNCLASSIFIED

F/G 28/11

NL





COORDINATES /ENTRIES	NODE	R	THETA	XL
463	.63	0		-.4564
464	.75	0		-.49
465	.63	0		-.09327705
466	.75	0		-.11162873
479	.63	180		-.1036
490	.75	180		-.07
467	.63	0		.218
468	.75	0		.218
469	.63	0		1.64381482
470	.75	0		1.61759259
471	.63	0		4.0
472	.75	0		4.0
473	.75001	0		1.61759041
589	.75001	180		1.94537256
474	.63	30		-.43276688
475	.63	60		-.3682
476	.63	90		-.28
477	.63	120		-.1918
478	.63	150		-.12723312
480	.63	210		-.12723312
481	.63	240		-.1918
482	.63	270		-.28
483	.63	300		-.3682
484	.63	330		-.43276688
485	.75	30		-.46186535
486	.75	60		-.385
487	.75	90		-.28
488	.75	120		-.175
489	.75	150		-.09813465
491	.75	210		-.09813465
492	.75	240		-.175
493	.75	270		-.28
494	.75	300		-.385
495	.75	330		-.46186535
584	.75001	30		1.63954765
585	.75001	60		1.69953594
586	.75001	90		1.78148148
587	.75001	120		1.86342702
588	.75001	150		1.92341531
590	.75001	210		1.92341531
591	.75001	240		1.86342702
592	.75001	270		1.78148148
593	.75001	300		1.69953594
594	.75001	330		1.63954765
496	.63	30		-.08036908
497	.63	60		-.04510387
498	.63	90		.0030693
499	.63	120		.05124247
500	.63	150		.08650768
501	.63	180		.09941564
502	.63	210		.08650768
503	.63	240		.05124247
504	.63	270		.0030693
505	.63	300		-.04510387
506	.63	330		-.08036908
507	.75	30		-.09626212
508	.75	60		-.05427972
509	.75	90		.0030693
510	.75	120		.06041831
511	.75	150		.10240071
512	.75	180		.11776732
513	.75	210		.10240071
514	.75	240		.06041831
515	.75	270		.0030693

516	.75	300	-.05427972
517	.75	330	-.09626212
540	.63	30	1.66225867
541	.63	60	1.71264813
542	.63	90	1.78148148
543	.63	120	1.85031484
544	.63	150	1.9007043
545	.63	180	1.91914817
546	.63	210	1.9007043
547	.63	240	1.85031484
548	.63	270	1.78148148
549	.63	300	1.71264813
550	.63	330	1.66225867
551	.75	30	1.63954951
552	.75	60	1.69953704
553	.75	90	1.78148148
554	.75	120	1.86342593
555	.75	150	1.92341346
556	.75	180	1.94537037
557	.75	210	1.92341346
558	.75	240	1.86342593
559	.75	270	1.78148148
560	.75	300	1.69953704
561	.75	330	1.63954951

*
 LINE NODES 463 463 474 475
 476 STEP 1 TO 484
 LINE NODES 464 464 485 486
 487 STEP 1 TO 495
 LINE NODES 465 465 496 497
 498 STEP 1 TO 506
 LINE NODES 466 466 507 508
 509 STEP 1 TO 517
 LINE CYLINDRICAL 467 467 6 1
 LINE CYLINDRICAL 468 468 6 1
 LINE NODES 469 469 540 541
 542 STEP 1 TO 550
 LINE NODES 470 470 551 552
 553 STEP 1 TO 561
 LINE CYLINDRICAL 471 471 6 1
 LINE CYLINDRICAL 472 472 6 1
 LINE NODES 473 473 584 585
 586 STEP 1 TO 594
 *THE MESH FOR THE OUTER TUBE IS GENERATED
 GVOLUME 472 472 471 471 470 470 469 469 EL1=6 EL2=1 EL3=4 NO=20
 GVOLUME 470 470 469 469 468 468 467 467 EL1=6 EL2=1 EL3=4 NO=20
 GVOLUME 468 468 467 467 466 466 465 465 EL1=6 EL2=1 EL3=1 NO=20
 GVOLUME 466 466 465 465 464 464 463 463 EL1=6 EL2=1 EL3=1 NO=20
 *THE COORDINATES FOR THE HEADER AROUND THE TUBES ARE DEFINED
 SYSTEM N=3 TYPE=CARTESIAN X=9.4 Y=0 Z=0
 COORDINATES / ENTRIES NODE X Y Z
 *INNER TUBE HEADER NODES
 925 0.0 0.0 0.0
 926 .156 0.0 0.0
 927 .218 0.0 0.0
 928 2.0 0.0 0.0
 929 0.0 0.0 1.125
 930 .156 0.0 1.125
 931 .218 0.0 1.125
 932 2.0 0.0 1.125

933	0.0	0.0	2.25
934	.156	0.0	2.25
935	.218	0.0	2.25
936	2.0	0.0	2.25
937	0.0	2.0	0.0
938	.156	2.0	0.0
939	.218	2.0	0.0
940	2.0	2.0	0.0
941	0.0	2.0	1.125
942	.156	2.0	1.125
943	.218	2.0	1.125
944	2.0	2.0	1.125
945	0.0	2.0	2.25
946	.156	2.0	2.25
947	.218	2.0	2.25
948	2.0	2.0	2.25

*NODES OF THE CENTERS OF THE TUBES FOR MESH GENERATION

949	0.0	1.0	1.125
950	.156	1.0	1.125
951	.218	1.0	1.125
952	2.0	1.0	1.125
953	0.0	3.0	1.125
954	.156	3.0	1.125
955	.218	3.0	1.125
956	2.0	3.0	1.125

*OUTER TUBE HEADER NODES

1230	1.41	4.7	2.25
1231	.218	4.2764467	2.25
1232	-.171	4.13822335	2.25
1233	-.56	4.0	2.25
1234	1.41	4.7	1.125
1235	.218	4.2764467	1.125
1236	-.171	4.13822335	1.125
1237	-.56	4.0	1.125
1238	1.41	4.7	0.0
1239	.218	4.2764467	0.0
1240	-.171	4.13822335	0.0
1241	-.56	4.0	0.0

*GENERATION OF THE MESH FOR THE INNER TUBE HEADER

LINE NODES 123 125 124

LINE ARC 68 70 NCEN=951 EL=1 M=1 NCO=ALL

LINE ARC 46 48 NCEN=950 EL=1 M=1 NCO=ALL

LINE ARC 24 26 NCEN=949 EL=1 M=1 NCO=ALL

GVOLUME 948 936 125 123 947 935 70 68 EL1=1 EL2=1 EL3=4 NC=N

GVOLUME 947 935 70 68 946 934 48 46 EL1=1 EL2=1 EL3=1 NC=A

GVOLUME 946 934 48 46 945 933 26 24 EL1=1 EL2=1 EL3=1 NC=A

*

LINE NODES 125 127 126

LINE ARC 70 72 NCEN=951 EL=1 M=1 NCO=ALL

LINE ARC 48 50 NCEN=950 EL=1 M=1 NCO=ALL

LINE ARC 26 28 NCEN=949 EL=1 M=1 NCO=ALL

GVOLUME 936 932 127 125 935 931 72 70 EL1=1 EL2=1 EL3=4 NC=N

GVOLUME 935 931 72 70 934 930 50 48 EL1=1 EL2=1 EL3=1 NC=A

GVOLUME 934 930 50 48 933 929 28 26 EL1=1 EL2=1 EL3=1 NC=A

*

LINE NODES 127 129 128

LINE ARC 72 74 NCEN=951 EL=1 M=1 NCO=ALL

LINE ARC 50 52 NCEN=950 EL=1 M=1 NCO=ALL

LINE ARC 28 30 NCEN=949 EL=1 M=1 NCO=ALL

GVOLUME 932 928 129 127 931 927 74 72 EL1=1 EL2=1 EL3=4 NC=N

GVOLUME 931 927 74 72 930 926 52 50 EL1=1 EL2=1 EL3=1 NC=A
 GVOLUME 930 926 52 50 929 925 30 28 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 129 131 130
 LINE ARC 74 76 NCEN=951 EL=1 M=1 NCO=ALL
 LINE ARC 52 54 NCEN=950 EL=1 M=1 NCO=ALL
 LINE ARC 30 32 NCEN=949 EL=1 M=1 NCO=ALL
 GVOLUME 928 940 131 129 927 939 76 74 EL1=1 EL2=1 EL3=4 NC=N
 GVOLUME 927 939 76 74 926 938 54 52 EL1=1 EL2=1 EL3=1 NC=A
 GVOLUME 926 938 54 52 925 937 32 30 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 131 11 132
 LINE ARC 76 6 NCEN=951 EL=1 M=1 NCO=ALL
 LINE ARC 54 4 NCEN=950 EL=1 M=1 NCO=ALL
 LINE ARC 32 2 NCEN=949 EL=1 M=1 NCO=ALL
 GVOLUME 940 944 11 131 939 943 6 76 EL1=1 EL2=1 EL3=4 NC=N
 GVOLUME 939 943 6 76 938 942 4 54 EL1=1 EL2=1 EL3=1 NC=A
 GVOLUME 938 942 4 54 937 941 2 32 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 11 123 122
 LINE ARC 6 68 NCEN=951 EL=1 M=1 NCO=ALL
 LINE ARC 4 46 NCEN=950 EL=1 M=1 NCO=ALL
 LINE ARC 2 24 NCEN=949 EL=1 M=1 NCO=ALL
 GVOLUME 944 948 123 11 943 947 68 6 EL1=1 EL2=1 EL3=4 NC=N
 GVOLUME 943 947 68 6 942 946 46 4 EL1=1 EL2=1 EL3=1 NC=A
 GVOLUME 942 946 46 4 941 945 24 2 EL1=1 EL2=1 EL3=1 NC=A
 *GENERATION OF THE MESH FOR THE OUTER TUBE HEADER
 LINE NODES 587 589 588
 LINE NODES 532 534 533
 LINE NODES 510 512 511
 LINE NODES 488 490 489
 GVOLUME 948 944 589 587 947 943 534 532 EL1=1 EL2=1 EL3=4 NC=N
 LINE NODES 532 510 891
 GVOLUME 947 943 534 532 946 942 512 510 EL1=1 EL2=1 EL3=1 NC=A
 LINE NODES 510 488 909
 GVOLUME 946 942 512 510 945 941 490 488 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 589 591 590
 LINE NODES 534 536 535
 LINE NODES 512 514 513
 LINE NODES 490 492 491
 GVOLUME 944 940 591 589 943 939 536 534 EL1=1 EL2=1 EL3=4 NC=N
 LINE NODES 536 514 893
 GVOLUME 943 939 536 534 942 938 514 512 EL1=1 EL2=1 EL3=1 NC=A
 LINE NODES 514 492 911
 GVOLUME 942 938 514 512 941 937 492 490 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 591 593 592
 LINE NODES 536 538 537
 LINE NODES 514 516 515
 LINE NODES 492 494 493
 GVOLUME 940 1238 593 591 939 1239 538 536 EL1=1 EL2=1 EL3=4 NC=N
 LINE NODES 538 516 894
 GVOLUME 939 1239 538 536 938 1240 516 514 EL1=1 EL2=1 EL3=1 NC=A
 LINE NODES 516 494 912
 GVOLUME 938 1240 516 514 937 1241 494 492 EL1=1 EL2=1 EL3=1 NC=A
 *
 LINE NODES 593 473 594
 LINE NODES 538 468 539
 LINE NODES 516 466 517


```

LINE NODES 494 464 495
GVOLUME 1238 1234 473 593 1239 1235 468 538 EL1=1 EL2=1 EL3=4 NC=N
GVOLUME 1239 1235 468 538 1240 1236 466 516 EL1=1 EL2=1 EL3=1 NC=A
GVOLUME 1240 1236 466 516 1241 1237 464 494 EL1=1 EL2=1 EL3=1 NC=A
*
LINE NODES 473 585 584
LINE NODES 468 530 529
LINE NODES 466 508 507
LINE NODES 464 486 485
GVOLUME 1234 1230 585 473 1235 1231 530 468 EL1=1 EL2=1 EL3=4 NC=N
LINE NODES 530 508 890
GVOLUME 1235 1231 530 468 1236 1232 508 466 EL1=1 EL2=1 EL3=1 NC=A
LINE NODES 508 486 908
GVOLUME 1236 1232 508 466 1237 1233 486 464 EL1=1 EL2=1 EL3=1 NC=A
*
LINE NODES 585 587 586
LINE NODES 530 532 531
LINE NODES 508 510 509
LINE NODES 486 488 487
GVOLUME 1230 948 587 585 1231 947 532 530 EL1=1 EL2=1 EL3=4 NC=N
GVOLUME 1231 947 532 530 1232 946 510 508 EL1=1 EL2=1 EL3=1 NC=A
GVOLUME 1232 946 510 508 1233 945 488 486 EL1=1 EL2=1 EL3=1 NC=A
*COORDINATES FOR THE REST OF THE HEADER ARE DEFINED
SYSTEM N=0
COORDINATES/ ENTRIES NODE X Y Z
1366 9.09 6.32 0
1367 9.09 6.32 2.25
1368 7.86 4.67 0
1369 7.86 4.67 2.25
1370 6.9 6.79 0
1371 6.9 6.79 2.25
1372 6.644 4.85 0
1373 6.644 4.85 2.25
1374 4.91 6.37 0
1375 4.91 6.37 2.25
1376 5.39 4.6 0
1377 5.39 4.6 2.25
1378 3.16 5.575 0
1379 3.16 5.575 2.25
1380 4.3 4.057 0
1381 4.3 4.057 2.25
1382 1.93 4.61 0
1383 1.93 4.61 2.25
1384 3.31 3.33 0
1385 3.31 3.33 2.25
1386 .77 3.03 0
1387 .77 3.03 2.25
1388 2.53 2.356 0
1389 2.53 2.356 2.25
1390 .17 1.55 0
1391 .17 1.55 2.25
1392 2.046 1.22 0
1393 2.046 1.22 2.25
1394 0 0 0
1395 0 0 2.25
1396 1.87 0 0
1397 1.87 0 2.25
1398 8.27 5.22 0
1399 8.27 5.22 2.25
1400 8.065 4.945 0

```

1401 8.065 4.945 2.25

*THE MESH FOR THE REST OF THE HEADER IS GENERATED

GVOLUME 1238 1230 1231 1239 1366 1367 1399 1398 EL1=2 EL2=4 EL3=1

GVOLUME 1239 1231 1232 1240 1398 1399 1401 1400 EL1=2 EL2=1 EL3=1

GVOLUME 1240 1232 1233 1241 1400 1401 1369 1368 EL1=2 EL2=1 EL3=1

GVOLUME 1367 1366 1370 1371 1369 1368 1372 1373 EL1=2 EL2=1 EL3=6

GVOLUME 1371 1370 1374 1375 1373 1372 1376 1377 EL1=2 EL2=1 EL3=6

GVOLUME 1375 1374 1378 1379 1377 1376 1380 1381 EL1=2 EL2=1 EL3=6

GVOLUME 1379 1378 1382 1383 1381 1380 1384 1385 EL1=2 EL2=1 EL3=6

GVOLUME 1383 1382 1386 1387 1385 1384 1388 1389 EL1=2 EL2=1 EL3=6

GVOLUME 1387 1386 1390 1391 1389 1388 1392 1393 EL1=2 EL2=1 EL3=6

GVOLUME 1391 1390 1394 1395 1393 1392 1396 1397 EL1=2 EL2=1 EL3=6

*THE NODES AT THE CENTER OF THE TUBES ARE FIXED

BOUNDARIES 111111 TYPE=NODES/949 STEP 1 TO 956

*THE NODE AT THE ORIGIN IS FIXED

BOUNDARIES 111111 TYPE=NODES/1892

*THE NODES ABOVE THE ORIGIN ARE ONLY ALLOWED TO ROLL IN THE X-DIRECTION

BOUNDARIES 011111 TYPE=NODES

932 1015 1018 1023 1026 1031 1034 1039 931 1043 930 1046 929

1895 1902 1906 1913 1917 1924 1928 1935 1939 1946 1950 1956

*THE NODES IN THE Y=0 PLANE ARE CONSTRAINED TO ROLL ONLY IN THAT PLANE

BOUNDARIES 010111 TYPE=NODES

936 1013 1049 928 961 1051 966 1017 1053 1054 973 1059 978 1025 1061 1062

985 1067 990 1033 1069 1070 997 1075 935 1041 1077 927 1004 1079 934 1044

1080 926 1009 1082 933 1047 1083 925 1394 1395 1891 1893 1894 1896 1900

1901 1903 1904 1905 1907 1911 1912 1914 1915 1916 1918 1922 1923 1925 1926

1927 1929 1933 1934 1936 1937 1938 1940 1944 1945 1947 1948 1949 1951 1396

1397 1955 1957

*THE NODES IN THE Z=0 PLANE ARE CONSTRAINED TO ROLL ONLY IN THAT PLANE

BOUNDARIES 001111 TYPE=NODES

1	2	3	4	5	6	7	8	9	10	11	17	28	39	50
61	72	83	94	105	116	127	133	136	139	142	145	148	151	157
163	166	169	175	181	184	187	190	193	199	205	208	211	217	223
226	229	232	235	241	247	250	253	259	265	268	271	274	277	280
283	286	289	292	295	301	307	310	313	319	325	328	331	334	337
343	349	352	355	361	367	370	373	376	379	385	391	394	397	403
409	412	415	418	421	424	427	430	433	436	439	442	445	448	451
454	457	460	463	464	465	466	467	468	469	470	471	472	473	479
490	501	512	523	534	545	556	567	578	589	595	598	601	604	607
610	613	619	625	628	631	637	643	646	649	652	655	661	667	670
673	679	685	688	691	694	697	703	709	712	715	721	727	730	733
736	739	742	745	748	751	754	757	763	769	772	775	781	787	790
793	796	799	805	811	814	817	823	829	832	835	838	841	847	853
856	859	865	871	874	877	880	883	886	889	892	895	898	901	904
907	910	913	916	919	922	941	942	943	944					
1014	1016	1020	1022	1024	1028	1030	1032	1036	1038	1040	1042			
1045	1048	1055	1063	1071	1122	1123	1124	1126	1128	1130	1131	1132	1134	1136
1138	1139	1140	1142	1144	1146	1147	1148	1150	1151	1153	1154	1156	1159	1163
1167	1174	1176	1179	1182	1184	1187	1190	1192	1195	1198	1200	1202	1204	1206
1210	1216	1222	1234	1235	1236	1237	1279	1280	1281	1283	1285	1287	1288	1289
1291	1293	1295	1296	1297	1299	1301	1303	1304	1305	1307	1308	1310	1311	1313
1320	1328	1336	1403	1406	1409	1412	1415	1418	1421	1425	1429	1433	1437	1441
1445	1448	1451	1454	1457	1460	1463	1466	1469	1472	1475	1478	1482	1486	1489
1493	1497	1500	1504	1508	1511	1515	1519	1522	1526	1530	1533	1536	1539	1542
1545	1548	1552	1556	1559	1563	1567	1570	1574	1578	1581	1585	1589	1592	1596
1600	1603	1606	1609	1612	1615	1618	1622	1626	1629	1633	1637	1640	1644	1648
1651	1655	1659	1662	1666	1670	1673	1676	1679	1682	1685	1688	1692	1696	1699
1703	1707	1710	1714	1718	1721	1725	1729	1732	1736	1740	1743	1746	1749	1752
1755	1758	1762	1766	1769	1773	1777	1780	1784	1788	1791	1795	1799	1802	1806
1810	1813	1816	1819	1822	1825	1828	1832	1836	1839	1843	1847	1850	1854	1858

1861 1865 1869 1872 1876 1880 1883 1886 1889 1896 1909 1920 1931 1942 1953
*THE INTERNAL PRESSURE LOAD IS APPLIED
LOADS ELEMENT

1 -2 700

STEP 1 TO

120 -2 700

55 -3 700

STEP 1 TO

60 -3 700

115 -3 700

STEP 1 TO

120 -3 700

126 -3 700

STEP 6 TO

192 -3 700

203 -2 700

204 -2 700

215 -3 700

STEP 12 TO

287 -3 700

216 -3 700

STEP 12 TO

288 -3 700

*

CONSTRAINTS

*THE FOLLOWING CONSTRAINTS ARE NECESSARY BECAUSE COORDINATES WERE

*DUPLICATED IN THE GENERATION OF THE MESH FOR THE HEADER AT THE

*BOUNDARY OF THE TUBES

968 1 1019 1

968 2 1019 2

968 3 1019 3

1020 1 1055 1

1020 2 1055 2

1056 1 1091 1

1056 2 1091 2

1056 3 1091 3

1127 1 1092 1

1127 2 1092 2

1127 3 1092 3

1159 1 1128 1

1159 2 1128 2

967 1 1160 1

967 2 1160 2

967 3 1160 3

980 1 1027 1

980 2 1027 2

980 3 1027 3

1028 1 1063 1

1028 2 1063 2

1064 1 1099 1

1064 2 1099 2

1064 3 1099 3

1135 1 1100 1

1135 2 1100 2

1135 3 1100 3

1163 1 1136 1

1163 2 1136 2

979 1 1164 1

979 2 1164 2

979 3 1164 3

992 1 1035 1
992 2 1035 2
992 3 1035 3
1036 1 1071 1
1036 2 1071 2
1072 1 1107 1
1072 2 1107 2
1072 3 1107 3
1143 1 1108 1
1143 2 1108 2
1143 3 1108 3
1167 1 1144 1
1167 2 1144 2
991 1 1168 1
991 2 1168 2
991 3 1168 3
1158 1 1177 1
1158 2 1177 2
1158 3 1177 3
1179 1 1210 1
1179 2 1210 2
1125 1 1209 1
1125 2 1209 2
1125 3 1209 3
1211 1 1248 1
1211 2 1248 2
1211 3 1248 3
1284 1 1249 1
1284 2 1249 2
1284 3 1249 3
1320 1 1285 1
1320 2 1285 2
1352 1 1321 1
1352 2 1321 2
1352 3 1321 3
1178 1 1353 1
1178 2 1353 2
1178 3 1353 3
1162 1 1185 1
1162 2 1185 2
1162 3 1185 3
1187 1 1216 1
1187 2 1216 2
1133 1 1215 1
1133 2 1215 2
1133 3 1215 3
1217 1 1256 1
1217 2 1256 2
1217 3 1256 3
1292 1 1257 1
1292 2 1257 2
1292 3 1257 3
1328 1 1293 1
1328 2 1293 2
1356 1 1329 1
1356 2 1329 2
1356 3 1329 3
1186 1 1357 1
1186 2 1357 2
1186 3 1357 3

1166 1 1193 1
 1166 2 1193 2
 1166 3 1193 3
 1195 1 1222 1
 1195 2 1222 2
 1141 1 1221 1
 1141 2 1221 2
 1141 3 1221 3
 1223 1 1264 1
 1223 2 1264 2
 1223 3 1264 3
 1300 1 1265 1
 1300 2 1265 2
 1300 3 1265 3
 1336 1 1301 1
 1336 2 1301 2
 1360 1 1337 1
 1360 2 1337 2
 1360 3 1337 3
 1194 1 1361 1
 1194 2 1361 2
 1194 3 1361 3

*THE FOLLOWING CONSTRAINTS DEFINE THE CONTACT SURFACES

585 2 552 2
 585 3 552 3
 586 2 553 2
 586 3 553 3
 587 2 554 2
 587 3 554 3
 588 2 555 2
 588 3 555 3
 589 2 556 2
 590 2 557 2
 590 3 557 3
 591 2 558 2
 591 3 558 3
 592 2 559 2
 592 3 559 3
 593 2 560 2
 593 3 560 3
 594 2 561 2
 594 3 561 3
 473 2 470 2
 584 2 551 2
 584 3 551 3
 1323 2 759 2
 1323 3 759 3
 1354 2 760 2
 1354 3 760 3
 1180 2 761 2
 1180 3 761 3
 1181 2 762 2
 1181 3 762 3
 1182 2 763 2
 1212 2 764 2
 1212 3 764 3
 1213 2 765 2
 1213 3 765 3
 1250 2 766 2
 1250 3 766 3

1251	2	767	2
1251	3	767	3
1286	2	768	2
1286	3	768	3
1287	2	757	2
1322	2	758	2
1322	3	758	3
1331	2	801	2
1331	3	801	3
1358	2	802	2
1358	3	802	3
1188	2	803	2
1188	3	803	3
1189	2	804	2
1189	3	804	3
1190	2	805	2
1218	2	806	2
1218	3	806	3
1219	2	807	2
1219	3	807	3
1258	2	808	2
1258	3	808	3
1259	2	809	2
1259	3	809	3
1294	2	810	2
1294	3	810	3
1295	2	799	2
1330	2	800	2
1330	3	800	3
1339	2	843	2
1339	3	843	3
1362	2	844	2
1362	3	844	3
1196	2	845	2
1196	3	845	3
1197	2	846	2
1197	3	846	3
1198	2	847	2
1224	2	848	2
1224	3	848	3
1225	2	849	2
1225	3	849	3
1266	2	850	2
1266	3	850	3
1267	2	851	2
1267	3	851	3
1302	2	852	2
1302	3	852	3
1303	2	841	2
1338	2	842	2
1338	3	842	3
123	2	90	2
123	3	90	3
124	2	91	2
124	3	91	3
125	2	92	2
125	3	92	3
126	2	93	2
126	3	93	3
127	2	94	2

128	2	95	2
128	3	95	3
129	2	96	2
129	3	96	3
130	2	97	2
130	3	97	3
131	2	98	2
131	3	98	3
132	2	99	2
132	3	99	3
11	2	8	2
122	2	89	2
122	3	89	3
969	2	297	2
969	3	297	3
970	2	298	2
970	3	298	3
971	2	299	2
971	3	299	3
1021	2	300	2
1021	3	300	3
1022	2	301	2
1057	2	302	2
1057	3	302	3
1058	2	303	2
1058	3	303	3
1093	2	304	2
1093	3	304	3
1094	2	305	2
1094	3	305	3
1129	2	306	2
1129	3	306	3
1130	2	295	2
1161	2	296	2
1161	3	296	3
981	2	339	2
981	3	339	3
982	2	340	2
982	3	340	3
983	2	341	2
983	3	341	3
1029	2	342	2
1029	3	342	3
1030	2	343	2
1065	2	344	2
1065	3	344	3
1066	2	345	2
1066	3	345	3
1101	2	346	2
1101	3	346	3
1102	2	347	2
1102	3	347	3
1137	2	348	2
1137	3	348	3
1138	2	337	2
1165	2	338	2
1165	3	338	3
993	2	381	2
993	3	381	3
994	2	382	2

994 3 382 3
 995 2 383 2
 995 3 383 3
 1037 2 384 2
 1037 3 384 3
 1038 2 385 2
 1073 2 386 2
 1073 3 386 3
 1074 2 387 2
 1074 3 387 3
 1109 2 388 2
 1109 3 388 3
 1110 2 389 2
 1110 3 389 3
 1145 2 390 2
 1145 3 390 3
 1146 2 379 2
 1169 2 380 2
 1169 3 380 3
 283 2 1124 2
 284 2 962 2
 284 3 962 3
 285 2 963 2
 285 3 963 3
 286 2 1016 2
 287 2 1052 2
 287 3 1052 3
 288 2 1088 2
 288 3 1088 3
 325 2 1132 2
 326 2 974 2
 326 3 974 3
 327 2 975 2
 327 3 975 3
 328 2 1024 2
 329 2 1060 2
 329 3 1060 3
 330 2 1096 2
 330 3 1096 3
 367 2 1140 2
 368 2 986 2
 368 3 986 3
 369 2 987 2
 369 3 987 3
 370 2 1032 2
 371 2 1068 2
 371 3 1068 3
 372 2 1104 2
 372 3 1104 3
 409 2 1148 2
 410 2 998 2
 410 3 998 3
 411 2 999 2
 411 3 999 3
 412 2 1040 2
 413 2 1076 2
 413 3 1076 3
 414 2 1112 2
 414 3 1112 3
 745 2 1281 2

746 2 1317 2
746 3 1317 3
747 2 1175 2
747 3 1175 3
748 2 1176 2
749 2 1208 2
749 3 1208 3
750 2 1245 2
750 3 1245 3
787 2 1289 2
788 2 1325 2
788 3 1325 3
789 2 1183 2
789 3 1183 3
790 2 1184 2
791 2 1214 2
791 3 1214 3
792 2 1253 2
792 3 1253 3
829 2 1297 2
830 2 1333 2
830 3 1333 3
831 2 1191 2
831 3 1191 3
832 2 1192 2
833 2 1220 2
833 3 1220 3
834 2 1261 2
834 3 1261 3
871 2 1305 2
872 2 1341 2
872 3 1341 3
873 2 1199 2
873 3 1199 3
874 2 1200 2
875 2 1226 2
875 3 1226 3
876 2 1269 2
876 3 1269 3
*PORTIONS OF THE MODEL ARE DEFINED FOR GRAPHICS
EZONE TUBE1
1 2 STEP 1 TO 60
EZONE TUBE2
61 62 STEP 1 TO 120
EZONE TUBE3
121 122 STEP 1 TO 156
EZONE TUBE4
157 158 STEP 1 TO 192
ZZONE COMB01 TUBE1 TUBE3
ZZONE COMB02 TUBE2 TUBE4
ZZONE COMB03 TUBE1 TUBE2 TUBE3 TUBE4
EZONE LOAD
126 STEP 6 TO 192 203 STEP 12 TO 287 204 STEP 12 TO 288
ZZONE LOADING TUBE1 TUBE2 LOAD

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